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AVIATION

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A REVIEW AND ANALYSIS OF THE *European Tour*

THE *Dornier Do. X*



IN TWO SECTIONS

Section One



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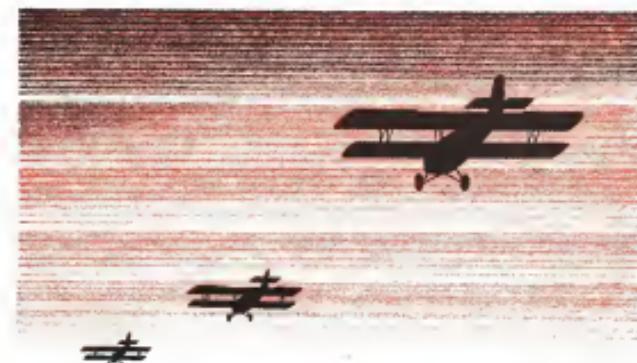
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When conditions are bad, all the aircraft converging on a city should be immediately under the command of a single official, with all the communication at his fingertips. There is no other way of safety. If radio beacon circuits have to be maintained on several parallel courses between the same cities leading to fields only a few miles apart, confusion will become intolerable.

We do not deny that there are communities in which the local circumstances are so peculiar that the regular use of several aerodromes, as opposed to the more emergency use of secondary fields when the regular one is covered with fog, may be desirable; but we believe that such cities are few. We believe that concentration is definitely in the general interest in most cases out of town. We speak for the passengers of the future and for air transport progress in expressing the hope that during the next few years at least, every possibility of agreement will be exhausted before more individual companies strike out on independent lines and without regard to their neighbors.



Seaplane Research

America is aseide to the seaplane at last. After years in which the voices of a few prophets crying in the wilderness echoed unanswered—years when even the Navy seemed content to allow the art of flying boat operation to perish along with the old F.E., and our need new designs and contributions to the aeronautical production of one or two new PNs at the Naval Aircraft Factory—flying boat fame has suddenly sprung up on every hand. Not able to wait for the evolution of Aeronauts designs in addition to the very few that have already proved themselves, promoters and financiers have crowded Europe for promising types. Already we have driven on the proscapes of Germany, France, Italy, and Great Britain, all of them either being built in the United States or planned for construction here in the near future.

All that is a consummation. It may seem too old a story to be worth repeating editorially, but it is easy to take it too much for granted. Assuming that we have a flying-boat industry, and that it is here to stay, what next? How shall we make it grow? We cannot resign ourselves to remain permanently patient followers in the footstep of Bleriot and Rohrbach, of Schenck and Short. We cannot rely permanently upon the constraining of commercial design to three parallelism with the more or less standard hull forms adopted by the Navy, excellent as is its record and great as is the credit that it sheds upon Captain Rohrbach and his co-workers in its evolution. We should restore our progress in a sorry measure if we denied to the partly commercial builder, having no direct relation with the naval service

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and seeking now, the opportunity of developing his own originality and displaying his own talents under the most favorable conditions.

Generally it is impossible, especially when the focusing of so much attention upon the giant seaplane, to depend on trial and error, or to hazard the whole fate of the new vessel in building a new type upon the efficacy and soundness of a novel form of hull based partly upon speculation.

To the fundamental question as to many others of the same type, there is only one answer: Research holds the key—and research upon seaplanes cannot be lasting, low, or water bound. If we let the aquatic parallel for the wind bound. By its use we are in position to determine quickly and cheaply and safely and accurately the performances of behavior of a hull upon the water, landing, or taking off. We can evolve at least empirical, if not rational, laws of designing for specified characteristics of action on the surface.

For nearly twenty years seaplane hull studies have gone on in Washington, handicapped by the limited space of maneuver possible in the Navy Yard, but by the inadequacy of instruments designed for a much lower class of work handicapped by pressure upon the capacity of experiment and staff and by the necessity of competing for a place on the schedule with studies of lethargy and destruction. Now, for the first time in this country, there is to be a laboratory having seaplane research as its first aim.

The aircraft industry does not yet seem to appreciate what cause for rejoicing there is in that fact. Those who see a great future for the seaplane and recognize the need of constant refined progress in the design of the type should fix an especially important attention on the National Advisory Committee for Aeronautics. They still will have some time to wait, but progress is being made. The day when the Committee opens the new high-speed wind tunnel at Langley Field, making it available for research and for the testing of the industry's own products, will exert in potential importance—we say it soberly and conservatively—any influence in American seaplane history in the last ten years.



Salesmen Not Pilots

IT IS AN AXIOM of selling that nothing can equal like-to-like personal contact. Selling certain types of mail order business, about all selling is done by word of mouth between two individuals, and much as direct mail and other forms of advertising may help, it is the personal appeal that wins success on the dotted line.

Aircraft selling is no exception to this rule and the need for more and better aircraft salesmen has been and is being felt. There has been much contention as to

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whether or not aircraft salesmen should first be pilots, the chief argument in favor of pilots salesmen being that only a pilot can speak the language of the air. On the other hand it must be admitted that more pilots-soldiers are pilots first and salesmen second. And in these days of increasing sales competition a salesman should be able to convince all his men on selling only.

It must be admitted that aviation as an industry did not advance greatly until business stepped in, and it seems now that sales organizations might profit by an infusion of fresh blood. After all, in selling strength and their services it is more important to talk to business men in their own language, that of dollars and cents than is in the conversation with aeronautical terms. More and more we must sell young men for aircraft and aviation sales work, not by putting them through flying schools and ground schools, but by organizing aviation business schools which will make a possible for these salesmen to intelligently discuss the economics of using the air as a medium of travel and commercial transportation.



Talking Safety

THE AIRCRAFT INDUSTRY'S problem is one of persuasion. Neither the manufacturers of aircraft nor the operators of air lines can retain control with their present patrons. Before we can consider that a sound footing has been established we shall have to multiply our personal contacts several times over.

There are today many thousands of people who do not travel by air, notwithstanding the progresses of simple means and simple motive. They are in a financial position not only to travel upon the air lines, but to satisfy their desire to buy planes of their own. Their time is valuable, and they spend much of it upon the road. Still they stay out of the air. Why? That is the question which we must answer.

There may be many partial answers, but the factor which certainly plays the leading part in an enormous number of cases is fear.

Some millions of Americans are still afraid to make regular use of the airplane as an ordinary vehicle of transport, and we shall do ourselves not a particle of good by blighting this fact or seeking to dismiss it from consideration by saying that all those so affected are fools or cowards, or that their state of mind is to be blamed on the newspapers or on some other third party. The attitude of those who ought to use aviation but do not is a fact. As such we must recognize it. As such we must vigorously contest with it.

The reputation of aviators to those who, because of lack of accurate information, or for any other reason do not agree with us that the airplane is safe enough

for ordinary employment without thought of hazard will get on nowhere. Those who do not feel disposed to patronize the air line cannot be shamed or brow beaten into doing so. Every intelligent man draws somewhere the line between what he considers justifiable and unjustifyable risks. Each of us will do certain things in a dire emergency that we would not consider proper as a matter of daily routine. We differ among ourselves only in the point at which our lines are drawn. What we have to do on behalf of aviation is to permit, by peace and example, by other argument and by emotional appeal, the great body of our fellow citizens that their individual fears concerning the activities which they consider proper for themselves should be driven to *absolute annihilation*.

We do violence to what many enthusiasts have reverenced as a basic article of their creed that safety should never be mentioned. They feel that it should be taken for granted, and that we should treat the passengers as though no humans on that score could ever enter their minds. Despite our respect for the courage and experience of those who hold that view, we are quite sure that they are entirely wrong. We cannot suppress this question by evading it. It seems to us rather that, if all discussion of safety be refused, the general non-aeronautical public will derive the impression that we have an inferiority complex on the subject and that the aeronautical people do not dare in talk of their safety record at length and in detail because they are ashamed of it. We do not for a moment believe that any such suspicion is justified, for the suspicion should be given no opportunity of arising.

It would suffice to be a rather sensible plan if we re-framed from taking so much stock in the much used phrase that the general public is aeronautically illiterate. From that, and it will take much educational work on the part of the entire American aeronautic industry before that pleasant and profitable state will be attained. There are two kinds of flying, safe and unsafe. At present the man in the street can not tell the difference. To educate him, and the members of his family, as to what safe flying is and in its everyday value is the major portion of our task.

This subject is too large to be covered in a single editorial. We shall return to it in some form and in various details again and again. For we are convinced that it is vitally important. We believe that the safety of flying under proper conditions should be argued freely. We believe that safety should be a central feature in aeronautical advertising and publicity, and that the facts should be plainly stated. Some of the propaganda which has gone out in the past has been so crude and obviously misleading as to reduce the intelligence of an intelligent reader. To distribute that sort of stuff is worse than to shun and evade the issue altogether, but there is a middle ground of intelligent and fearless discussion that is far better than either.



By CAPTAIN HARRY A. SUTTON
U. S. Navy

FIGURES A, B, C, AND D
show progress of down turn
considered at the NACA.
Illustrations on the same
model

THE PROBLEM of clearing an airplane when it will be efficient in normal operation and safe under the abnormal condition of a tail spin has rightly increased in importance within the last year during which a great many new airplane designs have been produced. Engineering has been rushed in order to permit early production of new models, with the result that, in many instances, serious difficulty has been encountered in meeting safety requirements. Substantial information on airplane spinning has been published to enable a good deal of progress to be made, but there are many interesting spinning characteristics of this in the primary object. The problem becomes more complicated when the aim is to produce an airplane which will be more efficient than existing passenger or mail carrying airplanes and which will recover from tail spins satisfactorily, because con-

ditions must be made in the design and their effect cannot always be predicted without extensive engineering investigation and experiment.

Studies made over a period of several years have demonstrated that there are several important factors which must be considered, principally, the effect of general control characteristics of the airplane, the longitudinal position of its center of gravity, its nose distribution, the variation of its lift characteristics near the angle of maximum lift, and stagger or overhang in a tail. It was evident at one time that these factors were not as susceptible to quantitative differences as others because of the absence of mutual interference effects between the wings of a tailplane, however this has proved incorrect in several instances. There are undoubtedly differences between monoplanes and biplanes in their

The subject of spinning characteristics of both military and commercial airplanes is of enormous interest at the present time. The analysis of these abnormal spins from which recovery is difficult or impossible is particularly important. It is, therefore, with special gratification that we present this article and the one immediately following, giving two different points of view, both based upon long practical experience in spinning tests and also in the derivation and application of theories

bearing upon the subject. Captain Harry A. Sutton, lately of the Material Division of the Air Corps, now of the technical staff of the Aviacon Corporation, will be recalled as having received the Distinguished Flying Cross, and also the MacKay Trophy for the most notable achievement of the year by a military aviator, for his work upon a military observation airplane which had shown dangerous spinning characteristics. He directed the experiments and made the test flights himself.



FIG. A



FIG. B

androtorsional motions during a spin, but these differences are not accurate controlling factors in determining the character of tail spins. Additional factors of generally lesser importance are gyroscopic torque, vertical position of center of gravity, engine operation, vertical fin surface distribution, shape and relative position of vertical and horizontal tail surfaces.

Rounds which will produce results generally applicable to all cases is made difficult by the numerous factors of wide differing character which must be investigated, each through a considerable range of possible influences, separately and in combination with the others. The actual motions of an airplane spin are not steady in a mathematical sense and are practically impossible to duplicate in a wind tunnel. By placing certain restrictions on the motion of a model, valuable information on autorotation has been secured, in wind tunnel tests however, these restrictions may be the cause of drawing erroneous conclusions. Any turbulent condition of airflow in wind testing has always caused difficulty in applying the results to full scale performance and autorotation tests made in a wind tunnel are not the answer. It is necessary therefore to use a rotating model in a tunnel. Douglas states in which models were released with an initial spin and allowed to fall freely for a considerable distance, produced some information of value, but the conditions of such tests are hard to control. It is difficult to provide sufficient height of free fall in undisturbed air and to prevent damage or change in alignment of the model. The effects of gust distribution are very difficult to simulate in wind testing, while the effects of gyroscopic torque and propeller slipstream are practically impossible to duplicate in small scale. Model testing in a series of a wind tunnel can be particularly valuable in determining relative efficiencies of various sizes and shapes of control surfaces, and in developing an efficient airoil which will have less tendency to autorotate than those now generally used.

The best design criteria now available for use in producing an airplane having safe spinning qualities were developed from study of full scale spins made on numerous airplanes of different types, combined with the application of previously known principles of aerodynamics and mechanics. In the further development of basic principles it is highly important that as much as possible of the experience gained in full scale spinning be made available for general study. Actual determination of the motion of spinning in full scale is exceedingly difficult; however, the continuous measurement of accelerations and pressure distributions at various points on the airplane during a spin is practicable and would provide valuable information. Tests of this nature are generally beyond the scope of commercial organizations and government research agencies must be relied upon for such investigations. The series of tests now in progress under direction of the NACA, should produce much information of practical value.

The lift of an airoil varies as a fairly uniform function with increase in angle of attack until near the angle of maximum lift, where the rate of increase in lift begins to drop off and the airoil is rendered unstable. In a wind tunnel at small angles of attack a considerable reduction in lift resulting from a change in air flow conditions about the wing. This is a fundamental characteristic of all airoils. In a normal straight stall in which the airplane does not turn or bank, the result is a gradual settling or a spin stop, depending on the elevator control available and the manner in which it is used. All pilots are familiar with this phenomenon. If the airplane is allowed to turn or roll while in a stalled attitude, the resulting motion causes a change in angle of attack and lift distribution along the wing span. The angle of attack is increased beyond the stalling angle on the downwind wing and is decreased on the upwind wing. This causes a decrease in lift on the low wing and a relative increase on the high wing which tends to increase the initial roll. The decrease in lift beyond the

still on the downward moving wing is accompanied by an increase in drag which tends to turn the airplane in the direction of the roll. The effects of a turn while the airplane is held in a stalled attitude may initial roll or turn increases unanticipatedly resulting in a spin. Most pilots are familiar with these characteristics and have found by experience the difficulty of stopping a roll or turn which has started during a stall without giving up gainful flying power by decreasing the angle of attack. Most of us have also found that a spin is relatively easy to lay down in initial stages. As the spin gathers momentum, other factors, such as inertia forces, gyroscopic effects, etc., enter in to make recovery difficult. These depend largely on the speed of rotation and the problem of recovery from spins can be considered as one of providing some means for stopping the rotation or of reducing the effective angle of attack to a point below the stall.

Considering in detail the various factors which determine the manner in which an airplane spins and the ease of recovery from a spin, it is natural that the controls be effective and easy to operate. Aileron control is the least important of the three controls after a spin has started, but poorly designed aileron controls are often responsible for inadvertent spins. This is due to yawing moments caused by their operation in maintaining lateral level, especially at low speed and high angle of attack. If a wing starts dropping at slow air speed and the ailerons are used to level the wings, the direction of downward moving air should be increased by that of the operating aileron on the opposite wing, so there will be no tendency for the airplane to turn. In many cases, the downward moving aileron exerts much greater drag than the one which moves up and the airplane tends to turn in the direction of the low wing. The use of rudder to prevent this often results in a stall which completely stalls the airplane and a spin results. This adverse operating moment due to ailerons can be eliminated by any one



Fig. 1



Fig. 2

of several well known methods, one of which is by use of differential aileron motion where the upward moving aileron moves through a greater angle than the one which is moving down. In general, the aileron is the first of the controls to lose effectiveness during a stall, and in a spin they are of small aid. Moreover, of the ailerons from one extreme to the other during a stall will often have a noticeable effect on the angle of attack and will give up a greater amount of drag-free airspeed than the elevator to take place, but their effect in stopping rotation is very small. If they were capable of exerting a large drag force at the wing tip some benefit would be secured, however this is contrary to the characteristics desired in aileron controls for normal flying operations.

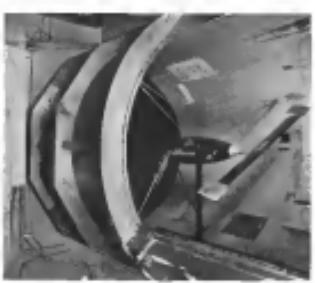
THE ELEVATOR and rudder controls are relied upon almost exclusively in recovery from spins and are generally used in combination, although in some cases it may be possible to stop a spin by use of either the rudder or elevator alone without regard to the position of the other control. The principal effect of the elevator is in controlling the attitude or angle of attack of the airplane. In starting a stall spin it is piled up in and the airplane and held there to bring it stalled. If the airplane is allowed to turn while thus completely stalled, either by use of the rudder or because of dropping a wing, the resultant roll will be a turn. A spin can be started by the use of the elevator alone with the rudder free but held in a neutral position, while it is practically impossible to start a spin by the use of the rudder alone if the elevator is left free or held on neutral. After the airplane has started spinning, the rotation sets up aerodynamic forces on the wings and, usually, inertia forces which tend to maintain a stalled attitude, and the spin will often continue if the elevator is released. The tail spin is one in which it is necessary to hold the elevator up to maintain the spin since in this case the airplane possesses an inherent tendency to dive out of the spin and will do so if the controls are released. As the spin continues,

if the elevator load is reversed and pressure is necessary to prevent the stick from passing back, the airplane will not stop spinning if the controls are freed and it may be necessary to exert considerable pressure on the stick to change the attitude sufficiently to stop the spin. In such cases the airplane will generally spin at a progressively flatter attitude if the elevator is left free until the force needed to decrease the angle of attack sufficiently to stop the spin may exceed that capable of being exerted by the elevator. Dependence most that is placed on the rudder to stop or slow up the rotation. The rudder is usually less effective in stopping a spin than the elevator. Sailing at the greatest cause of spinning and this is controlled by the elevator while the rudder exerts an effect in recovery. From a spin it is a common maneuver through doing up the rudder to recover the rotation. The effect of rudder from the stall stage is effected by the diving tendency of the airplane or by use of the elevator. It is to be expected in the normal case that the elevator would be more effective than the rudder in a spin. Both are generally used in starting a spin and in the recovery.

As the attitude of the airplane and its speed of rotation change during a spin, the airfoil conditions at the tail surfaces are modified and the effectiveness of the elevator and rudder may be greatly reduced. Tail surfaces are designed to be more effective under normal conditions when the air stream strikes them from the front at relatively small angles. In a spin the attitude of the airplane and the motions of rotation combine to greatly change the direction from which the air stream strikes the tail surfaces, so that it should not be surprising to find that at some point a particular control which is normally sufficiently effective is almost totally ineffective. Under these conditions the elevator is affected in the greatest, and in most cases, the rudder rotates effectiveness although the loads on it may be very high. No intrinsic design criterion exists for assessing this condition since no exact data is available regarding *in vivo*. It would seem best, however, to avoid rectangular stabilizers of large chord and rudders of small height above the stabilizer especially if they are entirely behind the stabilizer. Fig. 1 illustrates possible blunting effects. This is probably the reason that in some cases there is practically no air pressure on the rudder during a spin.

THE ANGULAR MOTION of elevators should be properly proportional as due the upward motion permitted is only slightly greater than the downward motion. Each should be capable sufficient for satisfactory control so that dynamic gain in maneuverability under the various load conditions, as in stall, control motion should be such that any normal pilot may, without difficulty, place the elevator in any conceivable position, particularly with the angle in the most forward position. The elevator control force should be small and the change in elevator force required for balance between power on and power off conditions should not be excessive. The elevator should retain effectiveness to the stalling point, while the rudder should be the last of the three controls to lose effectiveness in a stall and should have ample power at slow air speed without requiring excessive control force at any normal air speed. The rudder controls should be easily reached by a normal pilot with the rudder in either extreme position. Both control systems should be positive without appreciable slack or stretch.

Longitudinal position of the center of gravity affects springing characteristics in at least two ways through its



Wing twisted section for calculating loads made at NACA
Aerodynamics Laboratory, Langley Field, Va.

effect on inherent longitudinal stability of the airplane, and by the relative changes caused in mass distribution and inertia moments. An airplane which is made very stable by placing the center of gravity forward of its usual position is difficult to spin, due to the lack of sufficient elevator control to properly stall it, and to the tendency of the airplane to be very nose heavy during a spin. These effects are distinct advantages from the standpoint of safety from inadvertent spins provided sufficient longitudinal stability is provided for satisfactory flying characteristics. The wing loading airplane is balanced between 30 and 35 per cent of the mean aerodynamic chord and the longitudinal controls are required to provide sufficient effectiveness for sailing, making three power landings, and getting the tail up during takeoff at this range of biplane location. If the biplane on such an airplane is increased to 25 per cent of the M.A.C. it is generally necessary to provide more effective longitudinal control in order to stall the airplane or make satisfactory landings. A majority of the commercial airplane types have been so designed that the balance location with full load is between 35 and 40 per cent of the M.A.C. and in the cabin types particularly there is usually a large change in balance location between the full load and empty conditions. This is understandable both from the standpoint of control and longitudinal stability and has undoubtedly resulted from the fact that many airplane manufacturers recently is building thin designing airplanes. It requires careful care to design a biplane type airplane so as to insure that the longitudinal stability location will be satisfactory and on the other hand, with various load conditions will not be excessive, however, sufficient information is available and it is essential that it be applied if airplanes are to be made safe and otherwise satisfactory for commercial use. The small airplane requires that the balance be not further back than 25 per cent of the M.A.C. for satisfactory longitudinal stability, and for tail spinning it is highly desirable that it be nearer 25 per cent. Much has been written on the disadvantages of too much stability but these were more imaginary than real so far as longitudinal stability is concerned, and it is certain that many of the commercial airplanes recently designed in this

ometry lack the degree of longitudinal stability required for safe flight. To order to be satisfactory, an airplane must be designed for a limited range of balance location and large changes from this range cannot be made without risking corresponding changes in the design of horizontal tail surfaces.

All of the research on airplane spinning has clearly demonstrated that the center of gravity should be located



Fig. 1

well forward to ensure safe recovery from spin. In general, it has been possible to secure safe spinning characteristics with the balance as far back as 47 per cent of the M.A.C., but these are rare and were made possible because special consideration was given to tail controls and mass distribution in airplanes of unusual type. In most of the test article, as airplanes having unfavorable spinning characteristics it has been found that by moving the center of gravity forward to about 25 per cent of the M.A.C. satisfactory recovery from spin could be secured. This balance location would result in a high degree of longitudinal stability in the usual airplane and would insurmountable special problems in securing satisfactory longitudinal control.

In a particular case where the balance was programmed to move from approximately 19 per cent to 26 per cent of the M.A.C. in a staggered biplane, it was so difficult to get the airplane away from a spin that it was quite nose heavy during the spin. Recovery could be effected by releasing the controls, by placing the controls in the neutral position, by moving the stick all the way forward and leaving the rudder hard over in the direction of the spin, or by reversing the rudder and leaving the stick all the way back. The airplane was thus perfect so far as spinning was concerned, however, the longitudinal control was insufficient to overcome its inherent stability and permit three point landing to be made. Neither the stick nor the rudder was effective in this respect, but the use of ailerons had much effect on the spin. The airplane spun with its longitudinal axis inclined about 20 degrees from the vertical. As the balance was moved forward it became easier to get the airplane into a spin and it was necessary to use the control in recovery until, with the balance at about 25 per cent M.A.C. the airplane would not recover from a spin with the controls released. Recovery could easily be effected by reversing the rudder and elevator from the positions in which they were placed to start the spin. As the balance location was moved further aft it became progressively more difficult to stop the spin. Both the elevator and rudder forces became higher, the airplane spun with its longitudinal axis inclined at a larger angle to the vertical than previously, the spins were more irregular and slightly slower and it became necessary to use all of the

elevator and rudder control motions to recover from the spin. When a balance location of about 35 per cent of the M.A.C. was reached, recovery required holding the stick all the way forward, reversing the rudder and holding the controls in these positions for a considerable time against high control forces in order to stop the spin. Further forward movement of the center of gravity to about 39 per cent of the M.A.C. resulted in increase of control forces and time required for recovery to the point where it was absolutely necessary to hold the controls in their maximum reversed position for several turns of the spin before any tendency to recover could be noticed. Even a slight movement of the stick from its most forward position would cause the airplane to continue spinning. With that nose forward center of gravity location the airplane was longitudinally unstable and quite likely to roll over.

In the case of a staggered biplane in which the center of gravity was progressively moved from about 17 per cent to 39 per cent of the M.A.C. during spinning tests it was found that the effect on ease and certainty of recovery was approximately the same as in the staggered biplane, however there were great differences in the manner in which the two airplanes spun. The staggered biplane spun with its longitudinal axis inclined about 35 degrees to the vertical while balanced at 37 per cent M.A.C. and the spin became steeper as the center of gravity was moved forward. This airplane also spun slower under all conditions of balance although the change in speed of rotation with change in balance location was about the same in the two airplanes. Due to the uniqueness of attitude of spins with the most noseward balance location in the staggered biplane, recovery generally ended in a dive slightly beyond the vertical which would be an element of considerable danger near the ground.

In many other cases of tests on both biplanes and monoplanes the beneficial effect is a tilt of a forward

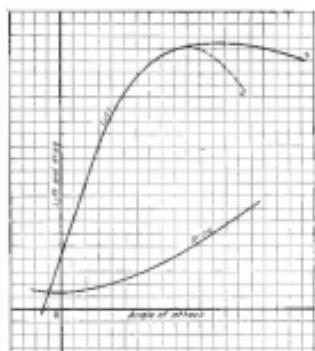


Fig. 2

location of the center of gravity has been demonstrated. The exact center of gravity location which will insure easy recovery from spin cannot be specified for all airplanes, however, it is apparent that, in general, where spin axis gyroscopic characteristics are unusual, the center of gravity should be located ahead of 30 per cent of the M.A.C. It is almost always easier to move the balance location rearward in a completed design than it is to move it forward.

In the case of each monoplane it is probable that center of gravity location is of more importance than in biplanes because the tail surface controls are affected

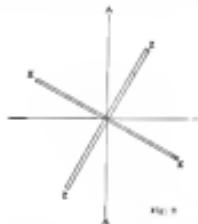


Fig. 3

more by the presence of the large closed body ahead of them and the airflow over the wing is subjected to greater disturbance. The effect of change in angle of attack of the airplane on tail surface controls is generally more pronounced in this type than in biplanes. Another factor of probable importance is the large vertical force component in a cabin airplane.

The effects of static distribution in an airplane on its spinning characteristics are confined mainly to the spin after it has been started. Although static distribution has its effect on the inherent tendency to spin through the effect of center of gravity location in longitudinal stability, in the majority of airplanes the moments of inertia are located in the fuselage. The moments of inertia about the three mutually perpendicular axes are then usually different and when the airplane rotates as in a spin, inertia forces are set up tending to bring the masses into equilibrium, which generally requires some change in attitude of the airplane. In the usual case where there is relatively little distribution of weight along the vertical axis and where there are large weights in the fuselage at a considerable distance from the center of gravity, the inertia forces resulting from a spin tend to cause the airplane spin flat or with the fuselage rotating in a horizontal plane. The effect of inertia forces is counteracted to some extent by aerodynamic forces and by the inherent tendency of fuselage to rotate. This is reflected when a fairly constant angle is maintained. Recovery from the spin requires that the elevators control be sufficiently powerful to overcome the moments of inertia as well as the aerodynamic forces tending to maintain the spin. Although moments of inertia of the airplane can be determined, it is difficult to determine the axis of rotation of a spin and the magnitude of the inertia forces involved. Rough approximations which have been made

in certain cases indicate that they may be very large especially in fast spins. Most pilots have experienced the effects of centrifugal forces in spins or in quick evasions and appreciate how quickly such forces increase under certain conditions and how great their effect can be in preventing the pilot from exerting the necessary force on controls to recover a normal flying attitude.

The tail center of mass distribution along the wing span has been investigated in a staggered and in a nonstaggered biplane for several balance locations. In the case of the staggered biplane it made very little difference in the spin or in case of recovery when large wing weights were placed on the wing at a considerable distance from the fuselage. The moment of inertia about the X and Z axes was increased by approximately 50 per cent over the initial values with relatively small increase in moment of inertia about the Y axis. The difference in spinning characteristics due to change in balance location was not appreciably affected by increased weight on the wings.

In contrast to this, the effect of placing additional weights on the wings of the unstaggered biplane was immediately apparent and caused considerably greater difficulty in recovery from a spin. The moment of inertia about the X and Z axes was increased by only approximately 20 per cent for two different balance locations before recovery from spin became very questionable. Although these tests indicate that positive stagger is desirable from the standpoint of safe spinning, it should not be assumed that these results would apply with equal force to all cases as the two airplanes differed in many other respects.

The limited experience which has been had in spinning two-engine airplanes with an engine on each side of the fuselage, has usually resulted in the pilot being very well satisfied with one such experience and, generally, recovery from these spins has been effected by using the static engine to assist in stopping the rotation. No exact data are necessary to show the desirability of keeping the moments of inertia of an airplane low by raising it considerably, but it is highly desirable that more information be secured on the effect of the relation between moments of inertia about the three axes. The following moments of inertia were determined for the two airplanes discussed above:

	Estimated Balances	Staggered Biplane
Y axis	0.40	0.40
Z axis	0.40	0.40
X axis	0.50	0.50

Calculations of moments of inertia from the detail design data, even though very carefully made, are subject to inaccuracy but may be made more reliable than experimental determinations unless these are carefully and accurately made. In general where a compound pendulum suspension is used, the length of suspensions should be short so great small errors in length of suspension introduce large errors in the moment of inertia determination. In a particular case, 0.1 per cent inaccuracy in the period and length of suspension caused errors in 7.68 per cent difference in moment of inertia. The maximum errors for a length of suspension of 57 inches and a period of 1.29 seconds resulted in a 2.9 per cent difference in moment of inertia.

Moments of inertia about the Z and X axes are the most important and it is the difference between these two moments of inertia about fixed axes in the airplane which causes difficulty. In Fig. 2, if the aircraft is placed in

rotation about the axis $A-A$, there will be a tendency for $Z-Z$ to rotate until it is perpendicular to the axis of rotation. This will occur unless $B-B$ is able to coincide with the axis of rotation $A-A$, but any slight displacement from this position will cause it to swing to a position at 90 degrees to $A-A$. The condition is thus one of unstable equilibrium. If the moments of inertia were reversed, $X-X$ would tend to rotate until it reached a position at 90 degrees to $A-A$. This applies in principle to the spinning airplane, although the axis of spin is not generally through the center of gravity but is at some intermediate angle, usually located on the inside of the spin, which makes the analysis of the problem somewhat less "squarely correct." In the same manner, $Z-Z$ will be forced to turn until $Z-Z$ and a dynamic couple will be created which will cause positive rotation or an increase in rolling attitude. It will be noted in the experimental determination of I_x and I_y given above, that there is little difference in the staggered location, especially if the calculated values are used. This may have accounted for the difference in their attitudes in a spin.

To assist in restoring safe spinning qualities, moments of inertia should be kept small by making the airplane compact, and particularly the difference between I_x and I_y should be small since the magnitude of inertial stability moments is determined by this difference, the attitude of the airplane, and the rate of rotation. In practice, any airplane in which the elevator control force reverses the roll moment, that is, where the roll of the aircraft when the elevator is zeroed and a horizontal attitude is assumed, has the moment of all the way back, should be brought out of the spin immediately. It should be noted that the effects of center of gravity location and inertia moments must be considered together, since changes in center of gravity location are usually made by shifting loads in the fuselage which also changes the moments of inertia.

IT IS WELL KNOWN that the fundamental cause of flat spins lies in the possibility of autorotation. The comparatively sudden reduction in lift force, or, more exactly, normal force on a wing past beyond its angle of maximum lift, is responsible for the possibility of a spin being automatically maintained after it is once started. This fact has been repeatedly demonstrated in theory and experiment, and it has been found that changes in shape of the lift curve, or in maximum lift, have a marked effect on autorotation. Instead of abruptly lifting off beyond the maximum, the lift curve could be made to remain at approximately its maximum through a considerable range of angle of attack, followed by a gradual reduction in lift forces as in Fig. 20. The probability of spinning would be greatly reduced. The relatively large angle of attack necessary for autorotation under such conditions would tend to eliminate spin in the normal airplane, since it would not have sufficient longitudinal control to reach such a large angle of attack. However, if a spin did get started it would probably be very difficult to stop due to the shuddering attitude. Most of the research on airfoils has been directed toward obtaining a higher maximum lift with low drag, and the most efficient airfoils exhibit a sharp peak in the lift curve. Various combinations of airfoils can be made to modify this feature but no airfoil combination of airfoils has as yet been developed which contains high lift, low drag, and the absence of a peak in the lift curve. A few airfoils using airfoils with comparatively flat top

lift curves have been built at a sacrifice in aerodynamic efficiency. Others have been built to accomplish the same purpose by providing longitudinal controls sufficient to permit lifting the airplane; however, this is obviously dangerous practice. Modification of wings by the use of wash out at the wing tips or by changes in plan form tend to delay lifting of the outer portion of the wing, but benefit mainly by decreasing the probability of inadvertent spinning and decreasing the rate of rotation during a spin.

THE USE of auxiliary surfaces or controls such as flaps and slots has been advanced as a means for spin recovery difficulties, and a large amount of experimental development has been carried on, particularly in England, where a large number of airplanes have been equipped with automatic leading edge slats. These are operated by the air force on the wing in such a manner that at large angles of attack they are opened thereby delaying the breakdown of smooth airflow over the wing and increasing the lift. At low angles of attack they are closed and conform to the contour of the airfoil section used so that aerodynamic characteristics at small angles of attack are practically unaffected. In a spin the inner wing operates at a higher angle of attack than the outer and the slot on the inner wing is presented to increase the lift and assist in recovery. Actually the angles of attack are sufficiently high at both extremes that generally both slats are open simultaneously. In the inner section of the slat, however, there is a slot to assist in preventing the air from lifting off the airfoil, thus increasing safety to some extent by generating a sense of security. By incorporating a locking feature which enables the pilot to bring into action only the slot on the inner wing during a spin, some aid in recovery may be expected, although it must be remembered that the slot also increases the wing drag. The development has not yet reached the point where a designer can be assured of exerting spinning difficulties by the use of slots.

The use of auxiliary devices intended to provide increased lift or drag on parts of the airplane where the results would be beneficial in recovery from spins is not desirable unless they can be made entirely automatic so that the pilot cannot be expected to operate any additional controls during a spin. It is to be hoped that no further addition will be necessary to the mechanics of an airplane because of them are needlessly complicated at present.

Present knowledge of spinning indicates that adequate and sound fundamentals in airplane design and the application of a few general principles should produce satisfactory results. The center of gravity should be kept well forward, about 30 per cent of the mean aerodynamic chord. This will also assist in securing longitudinal stability. The moments of inertia should be kept small, particularly the difference between moments of inertia about the vertical and longitudinal axes. All of the controls should be effective and easily operated under all normal conditions and especially at large angles of attack. Stagger should be used in a high-lift airfoil to effect better interference effect between leading and trailing surfaces in a roll about the centerline. It must be admitted that application of existing knowledge still leaves some element of doubt which only well founded research will dispel. Every engineer should be given competent research organizations engaged in solving this problem.

THE Flat SPIN

By LIEUT. RALPH A. OFSTE

U. S. NAVY

AS a general proposition it is true that any airplane which can be sharply stalled can also be stalled in spin. Further it may be readily seen that a large number of the smaller military and commercial aircraft which have been recently flown can be made to "flat" spin, although sometimes with difficulty and under somewhat unusual conditions as to landing, balance, and control disposition.

There seems to be but little agreement as to the proper terminology for this subject; in any one group of pilots it may be referred to as the inverted or inverted normal spin, autorotation, static spin, or uncontrollable spin. All of these, however, refer to one and the same thing—"flat" spin has been referred to cover the one because it is perhaps the most generally used although very flat spins are, strictly speaking, actually flat, use any they even approach that attitude.

It has been recently necessary to refer to the "flat" spin as a spin that is not completely inverted at the aircraft horizon, and a characteristic which moderately condemns the airplane in which it is encountered. It is quite true that a goodly number of pilots have lately been killed by crashing from prolonged spins which obviously were uncontrollable. It is not entirely probable that the flat spin has been with us just about as long as the conventional spin of years gone by, and that merely advancing knowledge and more spinning have brought up the newest determination of exact phenomena which differentiate the flat spin from the ordinary spins, namely. In any case the flat spin is definitely with us, and an endeavor is made in this paper to present some practical information relative thereto which has resulted from somewhat extended spinning experiments, and to offer certain criticisms of the present general attitude of the subject.

THE first trials of entry into a flat spin vary either widely between airplanes or may be markedly different as between two planes of the same size and loading. Generally speaking, however, the entry occurs as follows: the plane is stalled as in its normal spin usually at a level stall, but sometimes rather abruptly. Full air cleaner, and full rudder in spinning direction are retained thereafter. The plane is held in the conventional spin normal for the type, for from two to six or more turns. A normal airflow passes over the control surfaces, tending to move them toward the neutral position. This follows a brief period of transition, perhaps one

The present paper by Lieutenant Ofste and the preceding one by Captain Sutton admirably supplement each other, having been based upon studies of different types of planes and different methods of testing, and also treating the subject from somewhat different points of view. Captain Sutton dealt especially upon the general theory of spinning and the way in which changes in design affect spinning characteristics. Lieutenant Ofste concerns himself particularly with the practical operation of planes with abnormal spinning characteristics, and with the general place of the spin in present day aerial operation. Lieutenant Ofste, now attached to the flight test section at the Naval Air Station at Anacostia, has had a varied experience as a naval aviator.

turn, after which it will be observed that the rate of the spin has varied anywhere from a very small amount up to as much as 60 degrees, in the latter case being probably about on the horizon. The nose spin is lower than the outside, a peculiar whirling sound is noted, caused by the air stream striking streamlined wires and exposed slats at unusual angles, the engine vibrations as though the propeller were out of balance, the pilot may feel a brief period during the nose spin the seat by centrifugal force and rapidly, the control forces are reversed, let of stick and rudder and the seat will fall over and addition the inboard aileron (one moving inboard) takes charge to the extent of pulling the stick to the inboard corner. This is the "flat" spin, and unless some rapid action is taken the spin will probably remain flat and perfectly regular as an oscillation until the ground comes up and stops the whole proceeding.

ACTUALLY, what has happened? To start with, in the normal spin, the plane has been held in a stall by the elevators, over which a normal airflow has been passing. At the same time, the plane has rotated about an axis well beyond its normal sort of lateral descent with the wing surfaces at an average angle of attack only slightly beyond the bubble point of the airfoil. Recovery from this condition requires merely that the stall be discontinued (elevators moved down to reduce angle of attack and power greater forward speed), and that the turning motion be stopped with the rudder. Both of these operations are quickly effective because they strike the aerofoils at the right angle, and the airfoil is then given reduced air speed. But at these conditions occurs when the flat spin takes charge. The center of rotation has moved back to a point within the airplane, usually near the center of gravity, the angle of attack of the wings may now range from a small positive angle at the outer

up to 90 deg. near the center section and nearly 180 deg. at the main wing tip (that is, the main wing is moving backward with reference to the surrounding air), the elevators and rudder, instead of being held hard against a normal airflow by the pilot, are now held against the pilot by an unusual airflow; the engine thrust path, as now stated of being reasonably near the flight path, is now



The flat spin of a B-52, Fig. 2

almost at right angles to it, and the propeller tries to do its work at most unusual angles. These conditions all combine to cause the peculiar characteristics which the pilot recognizes as the flat spin.

The flat spin sounds rather bad, and sometimes it is. Certainly the first impression, at least in some types of planes, is decidedly disconcerting. But unfortunately the usual recovery takes the look for normal recovery, slightly exaggerated, and the stick is in the cockpit with open ailerons and rudder. The "feel" of the flat spin is still recover faster than two turns, some what about three, a few require a half dozen, and an occasional one simply will not respond to this treatment. The latter case is discussed later, the rule which holds in the very large majority of types, however, is to push the stick in the far corner, bear the rudder hard over, and pull them there.

To show the relatively wide variations possible in the spinning characteristics of different types the following examples are cited: the planes in all cases being biplanes, carrying full load, and all recovering from their normal spin in about a half turn:

(a) Two-place, high performance. Plane enters a normal spin quite readily. Makes about five extremely fast and regular turns with nose almost vertical, then slows down appreciably with nose rising about 20 deg. The normal spin is faster and much less uniform than the latter flat spin; recovery, however, requires two full turns from the latter condition.

(b) Training plane, 200 hp. class. Plane enters a normal spin quite difficult. The first three turns are slow and wobbly, with a distinct whip (about one turn) which subsequently raises the nose of the plane. After about the third or fourth whip the nose stays high, perhaps 30 deg. below the horizontal, and the speed of rotation almost doubles. Recovery takes a full three turns.

(c) Single seater, high performance. Apparently well

not flat spin unless stabilizer is adjusted full tail down. In the latter condition the nose quickly comes up to about 30 deg. from the vertical. Recovery very difficult, effected in two cases by pilots attempting to leave ship (with parachutes) and, on moving their bodies forward, ending the spin once and for all.

(d) Light two-seater, 100 hp. Spins easily, but with nose almost vertical. At about the fourth turn nose comes up about 30 deg., and flat spin is fully developed in the additional turns. Recovery is made in two turns.

(e) A training plane with 200 hp. engine. At a high pitch angle, with stick held at a small deflected angle, this craft spins with some difficulty. Once started, however, it goes into a flat spin after about three fast turns, and recovers readily in less than two turns. As a surprise, with the same deflected in the stabilizer, recovery takes three turns; but with stabilizer set for zero deflection the recovery requires about six turns.

The few examples cited above illustrate the rather wide variations in entry, altitude during spin, and time of recovery, that may be expected in a fairly diversified group of planes. But note the eyes are actually "flat," whether the nose be high or low, the same conditions obtain for all types of thrust, reversal of control forces, and markedly delayed recovery as compared with the normal spin.

A number of types seem to be borderline cases. A small increased angle of the elevators, a few degrees negative stabilizer, a few more pounds of load back of the center of gravity, and an apparently perfect surprise suddenly becomes a flat spin. Conversely, by setting the stabilizer positive, by keeping the load well forward,



Fig. 3

and by easing up on the stick just a small amount, an otherwise regular flat spin can be demonstrated to large numbers of the characteristics of this class. The name is surprisingly avoid in many cases.

The various air services permit prolonged spinning in several classes of small planes, with restrictions against certain specific types in which dangerous tendencies have been encountered. Some of the planes in service, particularly of the training types, have been so modified after their initial appearance that had spinning characteristics been less positively eliminated—at a cost, however, of some of the control and maneuverability, and with limitations on loading. The flat spin is definitely recognized, and, for all practical purposes, is accepted as bad news which must be taken with the good.

The regulations of the Department of Commerce diverge somewhat sharply from those of the military services. Their present requirements apply to open cockpit planes of less than 4,000 lb. and carry much of slightly less weight. Briefly they want that planes in these categories shall recover from a sustained spin in one and



one-half turns with retarded controls, no power, and no reversal of stick forces. While it is usually impossible to estimate the spinning characteristics of any plane, it is questionable if a number of approved commercial planes honestly do so. As has been previously stated, it is not difficult to pass a plane for test which due to a slight location on elevator throw, and moderate shift of axial loading will be incapable of entering a flat spin. Furthermore the pilot can without much difficulty avoid the flat condition by easing slightly on the stick, particularly in planes which have a pronounced "indicator" (swallowing of control stick) just before going flat. The point is to demand to know if a plane type might pass the Department of Commerce test but after getting out into flying service might develop so-called undesirable spinning characteristics, caused by slight change of load and control arrangements.

But after all, why should the flat spin be so horribly and generally condemned? Deliberate prolonged spinning is almost universally required before the airplane will go "flat." Just what purpose it is desired to accomplish, just what while the prolonged spin has, is



Fig. 4

difficult to see. Obviously every student pilot should have some spin training, but one or two consecutive turns are certainly sufficient to illustrate adequately the working of the spin and the recovery therefrom. These are different spins, mind you, at a safe altitude and with the pilot adequately warned as to what is about to happen. The characteristics of the spin are known, the student has been carefully instructed and drilled. He spins in order to learn what to do instinctively in case of an emergency spin.

Many accident-prone fashions are unfortunately placed in that category, that is, spins which originate close to the ground. Actually the spin or simply small dips off to one side with perhaps a half turn in that direction and goes on in an arc before being stabilized in a normal spin. Inherence of fatalities from this source involves proper training and immediate disposition of controls for recovery of flying speed. The point is that the plane does not even start spinning, and therefore this type of crash can have no relation to the flat spin.

The inevitable spin at altitude may result from any maneuver which leaves the plane in a stalled attitude. It may occur while maneuvering, proning turns or other maneuvers, or it may be caused by weather conditions in which a pilot, temporarily flying blind, has lost his sense of direction and ground relation. But once the plane has been stalled involuntarily, and falls to a spin, a qualified pilot should quickly recognize the condition and take prompt means for recovery. In the case of thick weather flying the pilot may do a succession of three

spins and spins; but each time the spin is recovered the man now has stick forward for recovery, and each time a dive is apparent (evidenced by high air speed) he uses his stick back. Of course it is not quite as simple as that, but unless the pilot is thoroughly experienced he would hardly be out attempting blind flying in dangerous weather. In any case no pilot will permit himself to stay in an involuntary spin much over a turn, and certainly, when the flat spin can be avoided by merely easing the stick forward he should encounter little if any trouble from this source.

Some pilots make use of prolonged spins in the testing of new types of aircraft, and particularly testing planes. Since it is not always done with necessary care, it is suggested that the man should follow the man who will turn him that danger and yet permit the correct data being obtained. The test pilot should first see that his plane is prepared exactly in accordance with production specifications. This applies particularly to the angle of attack of the elevators and rudder. The plane should



FIG. 2—Just before the crash. It will be noted that the plane has descended over a short period throughout the turn.

be flown from the most forward of the control envelope in order that centrifugal effect will be a maximum if a flat spin is encountered, the radius being that most characteristic of an axis passing approximately through the forward seat in the conventional double cockpit airplane. Finally a bigger or short duration should be rigged in the rear (cockpit) in which variable rates of small short can be phased during the progress of testing. The hopper is fitted with a needle damping valve with an operating handle close to the pilot.

Test spins should be started at an altitude certainly on the high side of 8000 ft., and once the plane has finished half a dozen turns of a normal spin, or a series for a few turns in a steady flat spin, recovery should be made. The normal recovery has already been described. If this proves ineffective after about six turns some more radical action must be resorted. Roll the stabilizer all the way forward. Bias my tail drogue, while retaining control in the maximum recovery position. Certainly in popular belief the engine is of distinct assistance in recovery from the flat spin, due to the resulting upwash aiding the tail surfaces more effectively and the propeller thrust aiding to give motion in the flying direction. Should this be ineffective after several more turns, a stronger effort should be made to rock the plane out, using the engine, reengaging each time the controls are moved for recovery. It is important of course to work with the natural period of the plane in attempting recovery by this means, the controls being operated very much in the same manner that a man plane is rolled on the step for takeoff. But if all the above recovery attempts prove useless the dump valve on the hopper is pulled, assuming that the plane has reached a loaded condition when trouble was encountered. This last should result in a movement forward of the center of gravity sufficient to permit normal recovery. If not, remember what paragraphs are made for.

Assuming that the plane has been prepared as previously outlined, the following specific procedure is suggested in the spin testing of new types of planes. The engine tests for speed, climb, stability, etc., will have been previously completed.

The first flight is made with center of gravity as far forward as possible, that is, with pilot in front seat and no ballast in rear. Set stabilizer for balance at cruising speed. With the plane without power, using maximum rudder and elevator. If plane goes into a single spin and refuses to spin, try again with full engine altitude and maximum stabilizer and elevator of the plane each time the plane will spin. If none of these is effective it is my suggestion that the center of gravity is too far forward to admit of recovering the plane in a stall.

The next flight should be made under the same conditions as the first except that an increment of shot load should be put in the hopper. On subsequent flights this load is gradually increased until the hopper is filled, and finally the full specification load is carried. In the latter case, if the flat spin has not yet developed an effect should be made to induce the condition as before by using full engine stabilizer and burns of the glands.

Finally if the plane does spin flat, and if it has dual control, a last resort should be made to determine whether recovery can be properly effected using the rear controls. For this purpose the shot hopper is removed, pilots flying in both seats with the forward one serving as safety pilot to take control in case the centrifugal force in the rear is too great to allow proper use of the controls plane.

But spinning is an uncomfortable, disagreeable, and frequently unhealthy. It is a maneuver unnecessary except for training operations. One can keep one's hand on an aileron by very occasional practice—not more than a couple of turns consecutively—on a training plane whose characteristics have been carefully determined. Beyond that as a general rule, adopt the policy of never spinning voluntarily. It gets you nothing except some dry perhaps, a good crack-up.

THE Dornier Do.X

A Flight Experience and Some Remarks on the Operation

By LIEUT. COMMANDER JAMES M. SHOEMAKER
U. S. Navy

SEPTEMBER fourth, 1929, was a proud day for the little town of Friedrichshafen on Lake Constance. In the morning the *Graf Zeppelin* sailed majestically up the Rhine and across the lake to its home harbor, where it was welcomed and applauded by thousands of gaily dressed and happy Germans headed by members of the Reich cabinet. The ambassador from the United States and from Japan were on hand to take part in the welcome and to review, on behalf of their respective countries, expressions of gratitude for the assistance afforded Dr. Eckener and his crew in the world-fight just completed.

After the crew and passengers of the *Graf Zeppelin*

Performance records of the *Do X* have been widely published, and many journalists have written extant impressions of the big flying boat. Lieutenant Commander Shoemaker, however, tells of his reaction to a flight over Lake Constance from the point of view of an expert, having been in Germany as official observer for the U. S. Navy on board the *Graf Zeppelin*. When he writes of the performance of the *Do X*, he is comparing it with a broad, scientific experience in all types and varieties of aircraft.

had been deposited at the Kurgarten Hotel, and were visiting their fest signatures for three days, a gradually increasing sense of many visitors was heard to the southwest. In a few moments Dr. Dornier's Interceptor, the twelve-cylindered *Do X*, sailing over-head, had glided down to a landing in the lake.

That same afternoon, the three American naval officers who had been guests aboard the *Graf Zeppelin* accompanied Ambassador Schuman and the official party in an inspection of, and flight in, the tremendous *Do X*. Two motor boats carried about fifty people out to the plane, which had been swinging at low mooring off the town. As we approached the plane, the



The giant Dornier biengined flying boat flying at anchor.

boats were dwarfed by comparison. The passengers were landed on the standard stub wing of the Do X, which was built by Dr. Dornier.

The party was ushered into the passenger space—a tremendous compartment in the hull. Shortly afterward the plane was cast off from its moorings and taxied slowly out onto the lake. The pilot tested out all engines for a moment, then heading into the light breeze, he

plane in. We settled on the water without the slightest shock, and descended gradually to a stop. The most striking feature of the Do X, aside from its huge bulk, was the ease of landing both on the water and in the air. Like the Bessone Super-Wal, the Do X has a water rudder, which is operated by a wheel alongside the pilot's seat. The plane was handled on the water very easily with only two motors running. As to controllability during take-off in the air and landing, the pilot seemed to exert no more effort than in handling a twin-engined flying boat. There are side-by-side dual controls, but the amateur pilot did not make them during take-off and landing. And the plane taxied off into a three-lane harbor in 35 seconds with a gross weight of approximately 40,000 kilos (88,000 lb.).

CONTRARY to what might be expected, there are no servo motors or other mechanical means for moving the controls. Dornier balances, similar to those used on the Wal and Super Wal, do most of the work. There are balances for each aileron and each of the elevators, and a balance on each side of the rudder. The elevator and the rudder balances are adjustable, the control surfaces themselves being non-adjustable. Balance adjustment is achieved by varying the height of the balances past a separate control wheel for elevators and rudder being located alongside the pilot's seat. It is noteworthy that the balances themselves are mounted on fixed vertical surfaces, and suspended by the control surfaces by linkages, and not from the main structural surfaces by linkages.

The use of twelve engines by the plane sounds like a matter of great complexity, but it worked out on the Do X, it is certainly simple and seemingly fool-proof. The plane has two throttles, one for each side of the plane, located together forward of his seat. The throttle rods lead back to the engine room behind the navigating compartment where the engines are run in or out of the pilot's throttle as necessary. There is a row of twelve lights in front of the pilot, one light for each engine. When an engine is disconnected from the pilot's throttle, this is indicated by the lighting of a lamp in the light box. The engine room crew keeps the engines synchroized, takes care of fuel and oil supply, watches oil pressure and temperatures, etc. The pilot handles his throttles just as he would in a twin-engine plane, his engine instruments being two tachometers.

The Do X carries a crew of ten men—two pilots, two engine room men, and an engineer for each of the six engines. These last-named men start the engines at the needles. Before take-off the men fire each man's oil line return to the engine room and reseal the engine room gang; the men in each of the four starboard rooms during flight.

It is astounding that the run from the twelve Siemens-Halske Jupiters is not climbing. The engines are not provided with exhaust collectors and mufflers, yet the noise is not at all excessive—no more than is normally present in any multi-cylinder airplane. One of the reasons for the absence of excessive noise is undoubtedly the low propeller speed. The engines are geared, and the propellers run at one-half crankshaft speed. Another reason is the use of thick-sectioned four-bladed wooden propellers. Four-bladed wooden propellers are used as all Dornier flying boats; their bi-plane effect giving smoother running engines.

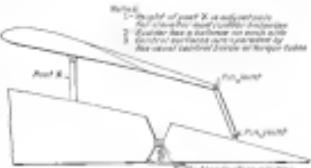
As to the Do X performance, Dr. Dornier stated that at the top speed of 212 k.p.h. (139.7 mph.) and the cruising speed was 180 k.p.h. (111.85 mph.). The landing and take-off speeds have not been measured, however, the plane took off in 60 seconds in a light breeze with a total weight of 40,000 kilos (88,000 lb.). The endurance at cruising speed, with a full load of fuel and oil and 100 passengers, is about six hours. So well had the design of the Do X been worked out that no trouble was experienced on the first real flight aside from the overheating of the rear engine. This difficulty was corrected by a cooling modification.

Most of the fuel and lubricating oil is carried in tanks in the hull (tanks are fabricated of riveted and welded sheet). Electric pumps are used to pump fuel and oil to the main tanks. Some difficulty may be experienced with the lubricating oil system in cold weather, due to the long lengths of oil piping. Electric indicator lights installed in the engine rooms show when the level of a tank is full. The tank is then pumped up more from the hull. An engine then pumps up more from the hull. The oil tanks are of the usual Dornier film type. The oil line

to the cooler is equipped with a thermostatically controlled valve, which cuts out the cooler when the engine is cold.

The original Dornier factory is located just west of Friedrichshafen, on the shore of the Bodensee (Lake Constance). At this factory, the Dornier planes are built. Due to the restrictions placed on German aircraft and engine construction by the Treaty of Versailles, Dr. Dornier established a new factory across the Bodensee at Altenrhein, Switzerland, for the construction of the Do X. The Treaty restrictions have been lifted, but the Altenrhein factory will continue to function. Planes as large as the Do X require factory bags the size of an airship hanger.

The Bodensee has its disadvantages as a flight-testing area for seaplanes. Due to the lake's altitude of 1,300 ft., with a consequent reduction in air density to only 90 percent of that at sea-level, unsupercharged engines cannot develop their sea-level horsepower. This reduction is horsepower due to altitude amounts to a total of 240 hp. for the Do X. Then, too, the fact that the Bodensee is



The interior arrangement of the Do X.

fresh water inhibits against the rate of takeoff of flying boats designed for salt-water operation. The Do X displaces two per cent more water in the Bodensee than she would in salt water, with a consequent marked increase in the thrust required to take off.

These disadvantages of the Bodensee aside the performance of the Do X is the more remarkable. When the big boat took off from the Bodensee in the light breeze it was about three feet above her capacity in salt water was removed, and with multiple engines providing a large factor of safety in the air, take off performance is the best record of the successful design of this remarkable flying boat.



The interior of the flying boat as 100 passengers were taken.

sped the thrupass wide. With many people on board, the plane fairly crawled out the "step" and 10 seconds from the time the throttles were opened the plane was in the air. We made a twenty minute flight along the shore of the lake, landing again in front of the town. The landing was a "Tut" site, with no attempt to stall the



The Do X took off carrying the great flying boat weekend passengers to the Bodensee.

Mapping 13,000 SQUARE MILES OF ALASKA

By LIEUT. E. F. BURKETT, U. S. N.
Executive Officer, Alaska Mapping Expedition

MAPPING of 13,000 sq. miles of northeast Alaska, not including 960 oblique photographs, tell part of the story of the 1929 Alaska Aerial Survey Detachment from Aircraft Squadrons, Battle Fleet. The survey covered principally the Togiak National Forest of which 4,000 sq. miles remain unmapped.

The photographic results of the expedition were 42 rolls of tri-lens images, 1,160, making a total of over 25,000 single photographs covering the area. The files have been delivered to the U. S. Geological Survey where they are filed for future use. In addition to these the services of 462 oblique photographs were turned over to the Forest Service. The Bureau of Aeronautics has on file the remaining negatives.

In addition to the value of these photographs in map making, they have proved of considerable value to the Forest Service. During the last two years prior to this during the 1926 expedition have been in the office of the Forest Service at Juneau, where officials have realized they are of great value in indicating the areas of the forests and to a large extent the quantity of timber in the areas over which the pictures were taken.

It is impossible to estimate, in dollars and cents, the value of the results of the expedition, recommended E. H. Stigges, representative of the Department of Interior and Agriculture, and former member of the expedition. "In fact, it will be years before the full worth of the results can be determined. Only those who have participated in the difficulties of conducting topographic mapping by ground methods in such a country as northern Alaska can appreciate the tremendous aid of the mapping pictures."

In this country, where climatic conditions are almost as they were, topographic mapping is slow, arduous and costly, not only because of the isolated weather but also because of the heavy blanket of timber and almost impenetrable undergrowth. Aerial topographic mapping by ground method is well-nigh impossible, at least within the bounds of watershed cost. At such methods, the topography of the shore lines and numerous ridges can be obtained with comparative ease, but accurate discovery and mapping of the valley bottoms with the many lakes and numerous streams is prohibitive. For this purpose the aerial photographs are invaluable and will serve to make the maps of the streams and

valleys of northeast Alaska much more accurate and possible of accomplishment in much less time."

But what did the survey undertake and accomplish? After the detachment was formed at San Diego and proceeded to Alaska aboard the *Gaspar* and a covered barge, and in the air on four Wasp-powered Loening Amphibians, the mapping planes flew 242,935 hours while undertaking mapping runs. Possibly the following data will be interesting:

Hours in the air for all planes, 27,40
Miles flown by all planes, 54,182
Gallons of gasoline used by all planes, 16,887
Hours in the air for photographic mapping, 242,935
Miles flown for photographic mapping, 19,636



One of the Loening Amphibians of the Alaska Survey Expedition in a snow-covered field.

Gallons of gasoline used for photographic mapping, 5,105
Hours of tri-lens mapping film exposed, 42
Number of tri-lens mapping exposures made, 2,380

Hours of "D" or fourth lens mapping film exposed, 10

Number of "D" lens mapping exposures, 1,500
Number of oblique mapping exposures, 25,000

Number of square miles covered by photographic mapping, 13,000

Number of prints made from mapping negatives, 600
Hours in the air for oblique photography, 95

Miles flown for oblique photography, 2,752

Gallons of gasoline used for oblique photography, 2,236

Number of aerial oblique exposure made

For Bureau of Aeronautics, 229
For other Federal Bureaus, 467

Number of prints made from oblique exposure, 3,600

THE PLANE flew overland from San Diego to Seattle enroute to their operating bases, but with few exceptions the regular flights made via the water passage was followed. The flight from Seattle to Ketchikan, due to the fact that it is nearly over well protected water and in case of a forced landing assistance would be more readily available.

As the distance between Seattle and Ketchikan could not be covered in one flight, Alert Bay, B. C. 204 nautical miles westward of Seattle in Johnson Strait was selected as the logical refueling point. All four planes of the detachment were taxied up on the beach. Although Alert Bay under present conditions is considered the most feasible stop-over point for planes flying between Seattle and Ketchikan, there are several other well protected harbors along this route where landing refueling facilities are available.

Similar to general conditions in southeastern Alaska

as concerned, later experience gave a clear-cut and positive proof of the adaptability of a rugged amphibious plane as the Loening for this work. No suitable land fields were available. Usually deep water and an extreme tide range of 24 ft. with resulting strong currents in the channels along which the most important cities are located usually made anchoring the planes impracticable.

At bases where sufficient dock space was available the planes were secured on the dock, the *Gaspar* having them to and from the water an operation required. Where dock space was not available, the aeronaut variable beach was selected and closed, the planes landing on the water and taxying out on the beach under their own power. It is in connection with such operations that the value of an amphibious plane is most apparent. The land bases were fairly large enough, the water covering up to the axles of the planes during the extreme high tides which occurred for a few days each month.

At Juneau, the *Gaspar* and barge were moored to the Government wharf and the planes secured on the wharf. This wharf was in excellent condition and equipped with telephone, water and power connections which were made available for the use of the Detachment. The operating conditions were very unsatisfactory and permitted only a few of the *Gaspar* because extremely efficient in handling the planes to and from the dock, and always gave excellent cooperation in the work to be done.

Before the summer was over, all fruit planes could be housed from the water and secured on the dock within 12 min. This became a vital factor in cutting down the interval at noon between mapping flights as the sun rose and the time limit for aerial photography was reduced. Aviation gasoline and oil in 50 gal drums was delivered to the dock by the Jettison branch of the Standard Oil Company of California.

Operating conditions at Ketchikan compared favorably with those at Juneau and photographic work was carried

red an from Kuchlau. It was decided to waive most of the art to be mapped. An aneroplane, with map equipment for taking upper air soundings and for making a daily weather map and forecast, proved its worth at this stage of the operations. Unfavorable weather for photographic flights was predicted for the next few days but on May 29 we had reached Peterburg and were ready to operate.

From Petersberg the islands of Kau, Coronation, Warren and the southern half of Barrow were mapped. After investigation and comparison of the facilities at Stikine and Juneau it was decided to shift the detachment from Petersberg to Juneau for the remainder of the work, so that the work could be completed in time for the opening of the Tonquin. On June 17 the entire detachment moved to Juneau, and while operations from this base the southern half of Barrow and the islands of Chichagof, Barrow and Yakutat and portions of the mainland were completed.

After opening from Juncos for two months, the detachment departed on August 17 for Kechikas arriving there two days later, after stopping at Pintuyary over Sunday in a dugout which contained without exception for seven days. When the weather finally cleared, the detachment mapped sections of the mountain between Bahia coral and Portada coral and a few scattered wreaths in the north along Stephens passage. Although there was much snow, a small portion of

Although we were under rain during the winter, the weather usually was the same throughout the area, although occasionally we were able to locate a clear area despite apparently unfavorable conditions. With this in view, it was important to arrange for weather reports at least once daily from as many local stations as possible. On one occasion, while based at Janzen, four successive mapping flights were made in the vicinity of Glacier Ranch, where it was cloudy, while at Janzen and to the south of the area was cloudy with occasional sun.

on with a minimum of lost motion. Aviation gasoline and oil in 50 gal drums was delivered to the dock by the Kachikan branch of the Standard Oil Company.

During the summer in southeastern Alaska such planes became the regular 20 hour check as required by existing instructions and it was largely due to the care and thoroughness with which these were carried out that engine failures were entirely eliminated. During rainy weather, which frequently occurred when the planes were due for this check a ~~cross~~ was rigged over the engine and the shock armrest, entering the roadway of the place for the next approaching weather.

Excellent performance was obtained from the four Pratt & Whitney air-cooled Wasp engines, and all four planes completed the summer's work without a change of engine. In view of the unfavorable operating conditions due to limited hangar space, lack of shelter, rapid temperature changes and heavy rainfall it is considered that both the planes and the engines withstand the rigors of not unusual weather.

Although beset with frequent and heavy seas, the detachment was ready to commence photographic operations on the day of our arrival at Ketchikan on May 28. Although photographic operations could have been car-



Surveying a region suitable for insurance. Policy research.



First: Shrubland; Forest: Shrubland and Liverworts: Shrubland (left to right, starting with fur outlined phasciaspines)



2000, and now, as the 1999 North Korean nuclear test, it is clear that

that going dead could not be determined. This necessitated carrying plenty of spare tubes in the plane.

clear enough for mapping early in the day, but cloudy, formed by noon. The first mapping weather after a day was usually extremely clear, but a day later the weather, even though cloudless, became slightly hazy with the haze increasing in density with the passage of each clear day. During our stay the following eighteen days were clear enough for mapping: May 20, June 4, 5, 6, 8, 12, 25, July 10, 11, 12, 18, August 1, 2, 3, 6, 7, 9, September 1, 5, 6. Of our 164 days in Alaska, rain fell on 73¹.

Three precautions were taken to insure the safety of any mapping crew. First, of course, the route of the flight was definitely determined in advance, with alternate flight lines in case of foul weather. Second, emergency rations were provided each plane to last in case of a forced landing at any distance from a base.

Flash plane was equipped with the standard map medical aviation first aid kit, and with a stock of emergency rations; an ordinary gear pack was, however, closed and soldered to make it entirely water tight. The rations included: 5 lbs can beans, 15 pieces hardtack, 2 lbs dried fruit, 2 lbs coffee, 3 lbs sugar, 2 cans salt, 2 packages meat, 2 pieces dried paper for making tea, 2 lbs powdered flour, 2 lbs ham, 2 lbs bacon, 2 lbs biscuits, 2 lbs cheese, 1 can of eggs, 1 can of sulphur matches, 2 lbs bacon and lime, 2 cans for making coffee.

Two of the four planes carried radio equipment. In making plans for the survey some difficulty was experienced in deciding just what aircraft radio equipment was desirable and necessary. It was finally decided a requirement was two SE 1367-5 transmitters and two SE 1367-6 receivers, and one SE 1376-II transceiver receiver. Two of the planes were wired for radio equipment before leaving San Diego, but radio sets were not carried between San Diego and Seattle.

Only a few minor radio troubles developed. Two fan developed vibrators and were replaced while the CW 1344 tubes after being in use a short while would become gassyified or lose their life. Care was taken to insure that the tubes operated at their normal filament voltage rating according to their tip coloring but the reason for

PRACTICALLY all flights required operation at altitudes well over 100 miles under atmospheric conditions generally very unfavorable. No high frequency set was available for the expedition. The excellent results obtained with the aircraft radio were due in a large measure to the efficient cooperation of the Canadian government, the U. S. Army and U. S. Navy radio stations.

While most important operations were carried on in the air, this could not have been accomplished without a base. Immediately after the 1939 expedition had been authorized, the Puerto Rican navy paid compensation for the use of the *YF-1* and the *YF-2* to the *YF-1* and *YF-2* respectively, a standard 110 ft by 30' very ammenable barge. It had been converted for use as living quarters, workshops, offices, store rooms and photography laboratory for the mission details of the 1936 expedition. The barge was equipped for hot and cold running water and was for electric lights and power.

The work during the summer was performed for several governmental departments, notably the Departments of Interior and Agriculture. Miss Condy A. W. Radford was officer in charge. Other officers and their responsibilities were as follows: Louis E. P. Hartman, executive officer, operations, navigation and communications; Louis F. R. Whitehead, photographic aeronautical; Louis F. G. Gubler, engineer skills, athletics; Louis (LP) L. P. Proffitt, assistant operations; Right, material; Louis Condy A. C. Smith (MDC) medical, welfare; Louis E. F. Carr (SC) supply, disbursing.

THE

Goodyear-Zeppelin Dock



Answer: A close up aerial view of the north face of the Bellvue Pinnacles. An air shot of the limestone bedrock, about 4000 feet thick.



10

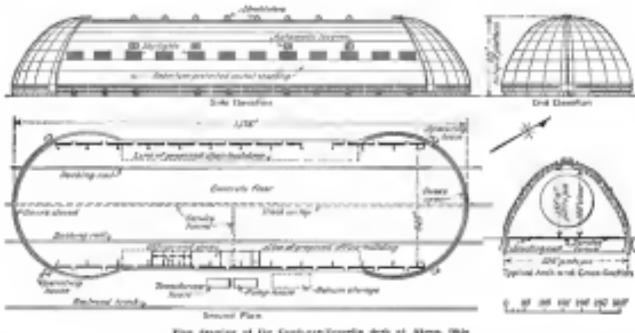
AS WORK PROGRESSES on the huge ship factory and dock being erected by the Goodyear-Zeppelin Corp. at the Akron (Ohio) Municipal Airport, it becomes more and more apparent that the structure is what its builders claim to be—the most massive and interesting building under construction anywhere in the world.

At the time of writing, the dock is approximately half finished. At an early date it should present a fairly complete appearance to the eyes of the operator. Work of assembling the first logie workshop inside the building is to start within a short time.

The 1958-9 and 1960-1, two super-Zigglers ordered by the United States Navy, will be constructed in the dock. It is highly probable that other air-liners—conventional ships—will be built there also. It is also likely that, eventually, another dock duplicate of the present one will be erected beside it. At least, the company is anticipating such expansion by providing ground enough

The two new Navy airships will be the largest ever built. They will be 785 ft. long and 135 ft. wide, and will hold 6,500,000 cu. ft. of helium gas. The dock is large enough to house a ship of 15,000,000 cu. ft. capacity.

If the new building were



Plan showing at the County courthouse site of Akron, Ohio.

the surface of each cross-section would be bounded by a parabola intersected by a straight line. If the shores were longitudinal and vertical, the water would be bounded by two half-parabolas connected by semicircular arcs. The overall shape is much like a huge lorgnette cone, eight lengths wide. The building is of large proportions. Its length is 252 ft., measured between centers of shore tracks; its width is 232 ft., center to center of each pair, and the height from the floor to the platform at the top is 211 ft. Area of the floor is 36,000 sq. ft., and cubic content of the building is about 34,000 cu. ft.

The Zeppelin shed built in 1906 at Friedrichshafen, Germany, is 650 ft long, 1308 ft wide, and 655 ft high. A later hangar built nearby is 787.2 ft long, 1508 ft wide, and 184 ft high. These, like most other European hangars, are of structural steel with sliding doors at the ends. At Orly, France, are two somewhat different types of sheds. They are of reinforced concrete, and are 984 ft long, 268 ft wide and 115 ft high. The larger hangar at Orly, France, is 850 ft long, 230 ft wide, and 115 ft high. It was built for the proposed English-Indian airship service. It is 850 ft long, 230 ft wide, and 115 ft clear height. The largest hangars in the United States other than the one at Akron are at Bellwood, Ill., and Lakewood, N. J. The Bellwood hangar is 800x150x150 ft. That at Lakewood is 800x200x172 ft. The ends of the hangars have vertical, sliding doors opening from the center outward.

In designing the Akron building, Dr. Karl Amstein, vice-president and chief engineer of the Goodyear-Ziegler Corp., had some problems upon which to work. Stellar structures, although on a considerably smaller scale, had been built and used in Dresden and Leipzig, Germany. A model in the general shape of the Akron building was constructed and put through rigorous wind-tunnel tests at the Dresdner Guggenbach School of Aeronautics.

securities of New York University. The model was 1/120 the size of the completed deck. Tests were conducted mainly by Dr. Aronoff and Dr. Wolfgang Klemperer, the latter also of the Goodyear-Zeppelin Corp. Working with the knowledge that wind, in striking a building and being deflected upward, creates a suction that tends to force the roof of the structure upward, they found that the deck would have to be braced against tremendous lateral forces from the inside. Section forces of this type sometimes are several times the magnitude of the windload. According to the code, Dr. Aronoff's experience was that the deck should be braced at 100 ft. intervals and not less than one-half the height, and that the roof should be secured against lateral forces equal to the maximum direct pressure. This practice of designing a building to withstand pressure from within has not been applied extensively in America, as seen in almost wholly negligible structures erected prior to the Akron deck.

MEASUREMENTS made at the Lubbock barge revealed that it is unsafe to dock an airtight when a cross-hull wind is blowing more than 10 mph. This velocity creates about the open, sliding-type doors, currents of twice the wind velocity, or 20 mph. Therefore it is evident that, if the dock structures can be lengthened to the same velocity as the cross wind, a 20-mile blow can be忍受ed, or an effective increase in usefulness of the dock by 100 per cent obtained.

It was this factor which led to adoption of the "spring seal" type doors for the new slack. They are designed so that a minimum of troublesome air currents is created, regardless of the door position.

Early in the ship's development it was learned that the hawser should be placed so that its longitudinal run is parallel to the prevailing wind direction. The Akers building is orientated to north 30° east by north 30° west, approximately.

Calculation of loads and stresses for the building was itself a huge task. The values finally arrived at include the following:

Dead loads—On arched: 30 lb. per sq. ft. On sheet rafters and purlins: 20 lb. per sq. ft. On lateral: 20 lb per sq. ft.

Snow loads—On all surfaces having an insulation of less than 45 per cent to horizontal: 30 lb. is maximum with dead load only, and 12 lb. in combination with snow.

Wind load—On all sheets, purlins and rafters, plus 30 lb. per sq. ft. for upper portion of building, and 40 lb. for lower. On all beams and arches, a wind pressure of 174 lb. per sq. ft. is allowed, and pressures are distributed according to previous discussions worked out from Dr. Aramian's experiments. A strong horizontal wind pressure of 15 lb. per sq. ft. on external surfaces only, with no consideration for internal forces, is to be used only when stresses due to this load exceed those from wind load.

Working stresses for dead and snow load have been set at 18,000 lb. per sq. in. on structural steel, and 24,000 lb. on silicon steel. For maximum stresses due to wind load, 24,000 lb. on structural steel and 32,000 lb. on silicon, are the values.

Carbon steel used in the building has an ultimate strength of 55,000 to 65,000, and a yield point of 30,000 minimum. Silicon steel has an ultimate strength of 80,000 to 95,000, and a yield point of 45,000 minimum.

General design of the building, finally, was selected because of its economy, simplicity and the minimum of structural weight.

This design, as developed by Wilbur Watson & Associates, consists of 11 parabolic arches spaced 80 ft. on center, and connected by a system of vertical and horizontal mass plates placed between the upper and lower chords of the arches, either than in the plane of the arch cords.

These trusses, in addition to forming the webwork of bracing for the outer shell of the dock, carry light tension cables spaced 30 ft. on center, and on these are placed 8-bar purlins 8 ft. on center. At each end of the main shell purlins are two diagonal arches, making the end and purlin system an 80 ft. span.

The outer purlins are held up in a similar way, of arches and braced ribs. All materials are of structural steel, except chords of the main arches which are manufactured with silicon steel.

Two of the outstanding features of the design are the novel type of doors used and the absence of expansion joints in the building. Of course, such a large all-steel structure will undergo a considerable change in overall size with fluctuation in temperature. Therefore,

the outer arches are fixed in position, while the remaining ones are carried on rollers. This permits the building to "stretch" and contract, much like a giant animal after a nap. The end arches, supporting the upper door busses, will move laterally about 4 in. under the maximum range of temperature. This motion is absorbed partly on the pins of the hinges and partly by the door structure.

If you can visualize one-eighth of an orange pad which stands 20 ft. high and is 214 ft. wide at the bottom, and is 10 ft. high at the upper portion of the top, you have an idea of the approximate size of the door opening. The curved buss road on 40 double flanged wheels rolling on curved tracks is fitted with the ground surface and resting on a concrete foundation supported by piles driven to bed rock. The weight of each section is 600 tons, or 1,200 tons for each end of the structure.

The supporting wheels are each of forged steel, 27 in. in diameter and the rolls upon which they run weigh 100 lb. per yard. The rolls are set to standard railroad gauge. To avoid chipping of wheels with consequent wear, the wheels are mounted individually, although positioned radially in pairs. Thus the outside wheels have to travel farther than the inner ones, can make more revolutions without interference from the others, as would be the case if every car's wheel were used with solid tires. The wheel bearing houses are again heavy forgings so that, if one wheel becomes more weight than its neighbor, the spring will give and tend to even the pressure. Since with a large mass of steel, it is almost impossible to obtain perfectly even distribution of weight, engineers have found.

Door sections revolve about large pins of the top of the dock. These pins are spaced 4 ft. apart, and are fastened to the roof girders by heavy steel frames. Each of the door girders exerts an vertical weight, but serves



All view showing the complicated ground operations between power supply of the new dock.



An aerial view of what the Goodyear-Zeppelin hangar looks like when completed. (Lockheed photo)

only to prevent the door from tripping inward or outward. Engineers have calculated that pressure against the pin will reach a maximum of 250,000 lb. inward, and 450,000 lb. outward, under the influence of snow and wind. Each of the pins is 12 in. in diameter and about 6 ft. long. The door sections are fastened to them through a bolt and washer joint, the half diameter being 50 in.

An interesting feature of the door is its porting to and from. It is built to permit several inches of the building to expand and contract under temperature changes. The bearing can move 4 in. up and down on the pins each way, to provide for thermal changes in the door height.

Moving such huge masses of metal as the door sections is no little task. In opening the doors, considerable power is required to start the sections in motion. Then, after they are moving, complete control must be possible at all times. Furthermore, the doors must not be permitted to move under wind action.

These conditions led to adoption of the "jack drive" system of door operation. Along the outside edge of each section, at the base, is a curved 14-in. H-beam; the radius of curvature being 192 ft. 10 in. Pins 4 in. in diameter are set along the beam in 9 in. pitch. The pins are engaged by the "huff" wheel of the driving mechanism.

The four driving units are located to enclose bases outside the door sections. They consist of a train of gears which operate a very large bell gear that engages in the door track. The gear train is driven, through a worm gear reduction, by an electric motor. The worm gears are externally self-locking, so that the door is held rigidly in a given position after the motor is stopped.

Each of the four door motors is built to operate at two speeds with a constant torque effect. It can drive the door at a speed of 25 or 40 ft. per min. The motors are General Electric, 200/100 hp., 15 min. rating, 600/300 r.p.m., 440 volt, three-phase, 60-cycle. A hydroelectric brake on the structure shaft serves to stop the motor, and is set to operate with a time lag of a few seconds after power is shut off, to permit the gears to slow down.

After the driving to open or close the doors, mostly power is used. Action from that point on is automatic. Electric contact switches set on the roadway next to the driving car are operated by projections from the floor. As the two leaves come together or approach the full open position, the switches are operated, and

the doors slow to a creeping speed, and ultimately stop in the proper position. The automatic feature makes possible the relatively high operating speed. In other hangars in this country, speeds of but 12 to 15 ft per min. are used.

Roller bearings are not used on the door wheels because it would then be too difficult to arrest motion, once it is started. It is true that such a wheel would permit easier starting, but the engineers decided to create a greater starting power rather than to complicate the problem of control when doors are closing down and stopping the sections.

It is estimated that an area of 20 tons pressure will be required to start the 600 ton doors in motion when snow and wind conditions are extreme, with the friction bearings used. Roller bearings also would not permit an high speed operation, because of the longer time necessary to stop the door motion. No trouble is anticipated from snow and ice getting onto the track and gear arrangement. The great pressure between the gear teeth and track pins readily dislodges such formations. The time required for opening or closing a door is estimated at 5 sec.

THE HANGAR is fundamentally a factory, and mechanical equipment is arranged accordingly. Handling equipment is limited to a single 10-ton lattice boom crane on a 120-in. 1-beam at the center of the building, overhanging. On either side of the center are two 10-ton lattice booms carrying working platforms which can be adjusted to desired height. Workmen will use these in assembling the airship. Other working platforms are located on 1-beams further down the curved building sides.

A fixed working platform extends from one end to the other of the building, in the center. On each side are six 1-beams, each 2 ft. wide, running the entire building length. They are located between the chords of the arches. Access to the upper platforms is by two stairways, one on each side near the center. There is also a specially-designed inclined railway for men and materials. Two counterbalanced cars run on a curved track bar in each section that each is horizontal at all times. The cars can be operated from the top of the building or from the roadway.

The building width between arch pins is 325 ft., but only 240 ft. of this space will be required for assembling the airship. The remaining area of about 42x60 ft. on each side will be taken up by shop and office equipment. At first shop will be dedicated to the

one allowing, in each category, twenty points for the fastest competitor, ten points for the second fastest and five points for the third fastest.

The aforementioned trials allowed no points for a speed under 52.8 mph, and 50 points for a speed of 89 mph and an intermediate scale for the first category, as regards the second category, the figures were no points for 43.5 mph, up to 50 points for 79.6 mph or above. In addition, competitors were dis-qualified for a speed less than 46.6 mph in the first category, and less than 37.4 mph in the second.

The final placings grouped the two categories according to the total number of points obtained.

Although somewhat arbitrary, it is safe to assume that it was the first time that light planes belonging to two different categories were to compete together and it was accordingly difficult to place perfect rules at the first attempt.

Nonstop of Caudron and Phœnix

THE AIR-CLUB of Czechoslovakia, France, Germany, Italy, Rumania, Switzerland. A notable absentee was Great Britain. But it was possible for a pilot whose National Association was not taking part in the Challenge to enter under the colors of a foreign Association of which he was personally a member. So British entries entered under the flag of the "Avio-Club de France" and others competed under German colors. In all, 82 entries were filed: four from Czechoslovakia, 28 from France, 34 from Germany, 14 from Italy, and 2 from Switzerland.

Modes of planes and engines may be summarized as follows: Czechoslovakia, four makes with Walter 80 hp engines; France, one make with Walter 80 hp engines; Phœnix, eight makes with Michelin, Anzani, Renault, Salmson, Hispano-Suiza and Gnome engines; Germany, twelve makes with Siemens, Argus, Gipsy, Salmson, Gnome, Cirrus, Wright, Goliath and Water engines; Great Britain, two makes, with Gipsy and Cirrus engines; Italy,



A "Phœnix" being towed through the "dust" during the racing time. Many cars were towing a load.

Due machine came in on Aug. 15, five Aug. 16 and

five makes with Clermont, Isotta-Fraschini and Fiat engines; Jagdwagen, two makes with Gipsy and Salmson engines.

Preliminary Trials

THE COMPETITION had to be at Orly Airport on Aug. 3, before noon, only for the opening of the competition. 35 of the 82 entries entered turned out on time and the weighing in procedure started. All ships complied with the maximum weight limit.

The machines were next inspected by the judges for the award of the "technical qualities" points. Out of a maximum possible total of 20 the best award was made in a Rovani 5, with 85 hp. First engine, receiving 18.25 points.

On Aug. 5, the competition test took place on the same Orly-AirArmy-Orly, to be covered twice non-stop (220 km.). The 35 competitors were dispatched by groups of four, the fastest machines first, and were sent out successively along the course. Seven were compelled to return owing to various mishaps, leaving 48 machines in the running qualifier to undertake the race itself.

Performance came to a great extent in the competition test. The best figure recorded was Nehring's (Darmstadt) D 18 65 hp Ganz) 18.5 points out of 20 possible, at a speed of 97.5 mph, 1st, next came Wirth (Klemm 25 40 hp Salmson), 12.5 points, speed 69.1 mph, 2nd, Roeder (Junkers A 95, 85 hp Ganz), Monfort (B 18 F 24 225, 20 hp Siemens) averaged 80.2 mph and V. Dangern (same plane and engine) averaged 78.4 mph.

The preliminary trials were now over; before the start of the race, the provisional placings were as follows: the total number of points possible being 46.

1. Kops (Avia 11, 85 hp Walter) 32.5
Wirth (Klemm 25, 40 hp Salmson) 32.5
Nehring (Darmstadt 18, 65 hp Ganz) 32.5
2. Lauer (Klemm 25, 40 hp Salmson) 30.6
3. Siebel (Klemm 25, 40 hp Salmson) 30.25
Roeder (Junkers A 95, 85 hp Ganz) 30.25

On Aug. 7, the machines were lined up on the field and started in groups of four at 3-second intervals, in their way to Orly. Weather was very bad and pilots encountered blinding rain and windstorms.

It would be too long and tedious to attempt recording the position of landing sites all along the circuit. The progress of various competitors soon became very confused since every one was entitled to choose at will the distance travelled each day.

At the conclusion of the preliminary trials Kleppen in Avia 11 with 85 hp Walter engine and Wirth, in a Klemm 25 with 40 hp Salmson engine, each had 32.5 out of a possible 46 points. Four other competitors had more than 30 points.

On Aug. 7, the machines were lined up on the field and started in groups of four at 3-second intervals, headed toward Orly. Weather was bad and blinding rain and windstorms were encountered. There was, naturally, a wide variation in speeds and progress along the way and it was impossible even to get an exact position at any point. But on Aug. 15, when the first check officials opened at Orly, some twenty planes were already landing at the airport waiting for the signal and, as the mud was deep, there was a dust toward the last in mass formation. A total of 25 planes arrived on that day before dusk.

Due machine came in on Aug. 15, five Aug. 16 and



A Junkers-D 1489, which made the fastest time to the race, taking off for the competition test. In the air is a Bleriot, which was made the best performance for endurance.

one Aug. 17, bringing the total number of survivors to 32, though one of them was subsequently disqualified for having failed to cover a leg of the race (160 km.) in 60 minutes, 20 planes figuring in the final rating.

The best performances in the first category were made by Cachery (Bleriot-Katzowen 25, 85 hp Ganz) and Caproni (Brodia, 85 hp Ganz) with 9.6 and 9.7 mph, respectively. In the second category, Monfort (B 18 F 24 225, 20 hp Siemens) averaged 80.2 mph and V. Dangern (same plane and engine) averaged 78.4 mph.

These four planes also had the best scores for the race, based on speed, reliability and number of spare parts. Cachery and Monfort had 119 points each, while Caproni Brodia had 109 and V. Dangern 106.5 points. The final rating of the contestants is not yet definitely determined, since the Italian government placed a premium against certain competitors who were suspected of having flown over prohibited areas. It is almost certain, however, that the following standing will be accepted.

[These results have been homologated since the article was written.—Ed.]

1. Wirth (Klemm, B 18 F 24 225, 20 hp Siemens)	100.5
2. Cachery (Bleriot-Katzowen 25, 85 hp Ganz)	100.5
3. Lauer (Klemm 25, 40 hp Salmson)	100.5
4. Ester (Bleriot-Katzowen 25, 85 hp Ganz)	100.5
5. Siebel (Klemm 25, 40 hp Salmson)	100.5
6. V. Dangern (Bleriot, 85 hp Ganz)	100.5
7. Caproni (Brodia, 85 hp Ganz)	100.5
8. Goliath (Klemm 25, 40 hp Salmson)	100.5
9. Monfort (B 18 F 24 225, 20 hp Siemens)	100.5
10. Goliath (Klemm 25, 40 hp Salmson)	100.5
11. Goliath (Klemm 25, 40 hp Salmson)	100.5
12. Goliath (Klemm 25, 40 hp Salmson)	100.5
13. Goliath (Klemm 25, 40 hp Salmson)	100.5
14. Goliath (Klemm 25, 40 hp Salmson)	100.5
15. Goliath (Klemm 25, 40 hp Salmson)	100.5
16. Goliath (Klemm 25, 40 hp Salmson)	100.5
17. Goliath (Klemm 25, 40 hp Salmson)	100.5
18. Goliath (Klemm 25, 40 hp Salmson)	100.5
19. Goliath (Klemm 25, 40 hp Salmson)	100.5
20. Goliath (Klemm 25, 40 hp Salmson)	100.5

The first "Challenge de l'Automobile" went heavily against the biplane: 58 out of 82 entries (70.7 per cent) and 28 out of 31 finishers (90.4 per cent). Of these 28, 15 (55.6 per cent) were pure contestants, the remaining 13 being of the rugged biplane type. As regards long duration, 15 (53.6 per cent) won long, the rest belonging to the "parasol" type.

The construction was in the majority of cases wood and fabric with a few fuselages of welded steel. Some machines had only a single landing gear. There was, naturally, only a "normal" construction, the "Junkers" which were of duralumin throughout according to the well-known practice of the firm. Despite that, however, the planes were not as heavy as might be supposed, with 85 hp engines the weight did not exceed 725 kg. empty, very translatable otherwise. But it is doubtful if this form of construction possesses marked superiority in the case of a light plane. Duralumin is extremely brittle, but repairs are much more difficult as skilled metal workers trained in the handling of duralumin are to be found very rarely. On the contrary, a machine built of wood throughout is nearly as robust and, in the case of damage, both easier and faster to be made ready.

It must be remarked that the folding or dismantling test was not compulsory, in fact, a certain number of competitors deliberately ignored it, thereby losing some

actual contest, the first arriving too late and the second lacking on the air whilst performing maintenance. The Darmstadt was compelled to land during the Tour and was unable to take part in the race, as the engine stopped. One Raud-Katzowen only completed the trial.

It was a disappointment not to see the two Cervia Aviators, their performance would have been watched with interest. None of them could be got ready in time.

It is significant that machines of the second category (less than 880 kg. empty) were much less numerous, only 18 out of 82 entries, and 8 out of 31 finishers. Such machines appear to be more suited for per-sonal and short journeys than for rail touring. In the hands of exceptionally good pilots, they are able to hold their own even in severe conditions, but their use for private owners is somewhat restricted. Except in very special cases, the weight limit does not allow of an engine of more than about 40 hp to be installed; this appears too few to enable two people and a sufficient amount of luggage to be carried in a reasonable average through any weather. The engines are open, which does not provide enough insulation. On the other hand, the appearance of passengers differed largely according to the type of cockpit and how it was arranged. Some planes had a closed cockpit, others were open and showed signs of severe use; some samples of cracked fairings. On the other hand, the covers of cabin doors emerged gaudily and remarkably fresh. Such facts mean a great deal for popularization.

In the larger class (less than 991 kg. empty), the power varied between 70 and 95 hp, planes were much more sturdy and comfortable, handles being appreciably faster. Some of the entries were extremely roomy and well designed, allowing for side seats to be installed and leaving large space for luggage and spares. The other hand, the covers of cabin doors emerged gaudily and remarkably fresh. Such facts mean a great deal for popularization.

In the larger class (less than 991 kg. empty), the power varied between 70 and 95 hp, planes were much

Technical Remarks

THE MONOPLANE scored heavily against the biplane: 15 out of 82 entries (70.7 per cent) and 28 out of 31 finishers (90.4 per cent). Of these 28, 15 (55.6 per cent) were pure contestants, the remaining 13 being of the rugged biplane type. As regards long duration, 15 (53.6 per cent) won long, the rest belonging to the "parasol" type.

The construction was in the majority of cases wood and fabric with a few fuselages of welded steel. Some machines had only a single landing gear. There was, naturally, only a "normal" construction, the "Junkers" which were of duralumin throughout according to the well-known practice of the firm. Despite that, however, the planes were not as heavy as might be supposed, with 85 hp engines the weight did not exceed 725 kg. empty, very translatable otherwise. But it is doubtful if this form of construction possesses marked superiority in the case of a light plane. Duralumin is extremely brittle, but repairs are much more difficult as skilled metal workers trained in the handling of duralumin are to be found very rarely. On the contrary, a machine built of wood throughout is nearly as robust and, in the case of damage, both easier and faster to be made ready.

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GENERAL NEWS

Introduce Aero
Loan Bill in HouseOther Aviation Drafts
Presented by C. J. McLeod

WASHINGTON (UPI)—Providing for the establishment of commercial aviation, the extent of a government loan fund of \$100,000,000, and a bill introduced into the House of Representatives by Clarence J. McLeod of Michigan, has been referred to the Committee on Interstate and Foreign Commerce.

Introduced as the Committee has a crowded docket, with such pressing measures as the Senate bill to be referred to the Senate by the House, the bill to expedite of railroad consolidation, it is not probable that Mr. McLeod's bill will receive attention for some time to come. The bill itself is one of 16 such bills introduced by the House to expedite the transportation industry in general. But a thorough study of it by the committee is indicated, which will serve further to elucidate its appearance on the floor of the House.

Provisions are made in the bill for the establishment of a loan fund not to exceed \$100,000,000, the use of which is to be extended to individuals or corporations engaged in aerial transportation or in the manufacture of aircraft to be received by them on aerial freighters and "other necessary facilities to the success of commercial aviation," and is to be repaid in full at the end of seven years.

Taxes Are Drafted

Friction over the use of the loan is due at the present time, as Mr. McLeod further provides that loans can not be contracted more than one-half the value of strength held or so to be held, or bought, or otherwise held, before the bill is introduced by the Department of Commerce or a board selected by the Secretary of Commerce of members, at least 75 per cent of whom must be experienced and qualified in the operation of aircraft.

One of the most important clauses of the bill is that requiring all strength, hangars, and facilities of individuals or corporations providing the loans to be placed at the disposal of the President of the United States in time of national emergency.

The point has been made in editorial quarters that this can only be accomplished by the use of the Post Office, as the Post Office air mail facilities, which would assist in the ventilation and support of commercial aviation, while rendering a more complete service to the public, and developing a trained

personnel for national defense. It has been suggested that, owing to the "rapid development of aircraft and aircraft technology," the government's policy would not be sufficiently protected.

Two other bills were presented by Mr. McLeod at the same time. One provides for the establishment of a permanent committee which would conduct the work now done by the Aeronautics Branch under the Aeronautics Committee.

To Use Army Craft
In Highway Project

SOUTHERN BELL (cont.)—Proposed Southern Airports Committee from President Franklin D. Roosevelt will be used on feasibility. The road would come for the relocation of the George Bush Highway, from Interstate 40, and the Latin South, the rough and hilly mountainous terrain that the new road will traverse makes the survey particularly adaptable to the use of planes.

of Commerce. The new division would be headed by a secretary, making self-sufficient members and he would appoint a director of aeronautics.

As far as possible, the new division and its subcommittees would be created to "foster, promote, regulate, and develop the science of aeronautics, and the adoption of flying, both of private, business and lighter-than-air craft to commercial purposes."

The third bill concerns a standing House committee on aeronautics, which would propose all proposed legislation relating to civil and military aviation. This group would be composed of seven members.

Firestone Fund on Tour

AKRON (cont.)—Plated by William M. McDonald and Edward J. Daigler and carrying four passengers, the plane came to the Akron plant, one of the principal aircraft centers of the country. There, representing the three-year tour of the West Coast states and the South are H. M. McElroy, president of the chamber of commerce; R. B. Patterson, manager; R. J. Bauman, manager of the plant; and Robert H. Beckwith, secretary and treasurer of the *Business Community Newspaper*. Last June the plane made a stop at Miami, Florida, on its way to the Miami Aircraft Meet in October.

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Florida State Aero
Association Formed

TALLAHASSEE (UPI)—Organization of the Florida State Aeronautics Association will be effected here in a meeting of aviation officials and citizens held at the call of Governor Carlson to discuss ways and means of developing the aviation industry in the state. Wallace D. Tamm, was selected president over Fred C. St. Peterburg, secretary. The first meeting of the association will be held at Miami on Jan. 18, with the opening of the Third All-American Air Meet. Gov. Carlson, who is president of the executive committee and the following executive committee was appointed: Adel A. W. Marshall, head of the United States Naval Air school at Pensacola; C. R. Shultz, Orlando; Dr. Roy C. Gross, Jacksonville; C. L. Whaler, Tallahassee; Frank Booth, Clearwater, and J. E. Vugt, Miami.

The executive committee was directed to conclude arrangements for the meeting, to select a chairman and a local representative, to arrange a meeting, create stimulation of interest, elect delegates to the gathering, and to make ready a constitution and by-laws.

The organization will have its headquarters at Tallahassee. It has as its purpose not only development of aviation, but also the furthering of branches of that industry into a single unit, the promotion of flying and to establish Florida as an air port to South and Central America.

The society has pledged itself to work for the establishment of emergency landing fields in the state and preparation of aeronautics for the use of the legislature charged to develop the organization which will be taken up at the future meetings.

Delivering Several Silverskys

BRIDGEPORT (cont.)—Silversky officials announced delivery of the sixth aircraft to the New York City Air Mail Air Lines, which will be put into service on the Miami-Havana leg of the international route. An application has also been received by Gentry Graves Co. makers of Maypole Radios, Inc., for the use of the aircraft. The Silversky organization for the Latin-American Airlines, Honolulu, H. I., has been granted delivery to that company.

Plan Smaller Ryan Plane

ST. LOUIS (UPI)—"High sister to the 'Spirit of St. Louis,'" will be in production here shortly after the first of the year, a Ryan Aeronautic Corp. announcement said.

Ryan Aircraft Plans
Extensive Sales System

ST. LOUIS (UPI)—Introducing a new policy announced by the Ryan Aircraft Corp. division of D.A.C., Ryan planes will be on sale in 100 cities in the United States, of a population of 10,000,000, at least as far as possible, so that suitable airports may eventually be offered to purchasers of somewhat lower priced," Mr. Malone said in conclusion.

method not far different from those used in advertising other products, and we believe that a sales organization of this nature, being as far as possible, so that suitable airports may eventually be offered to purchasers of somewhat lower priced," Mr. Malone said in conclusion.

Air Heads Express
Opinions to President

WASHINGTON (UPI)—Col. Paul F. Henderson, president of National Air Transport, and M. K. Keys, president of American Air Transport, Inc., on recently given President Eisenhower their views on the aviation industry situation and outlook. Col. Henderson declared the air mail business was good, passenger transport acreage only fair, and the future of the industry appeared to be somewhat brighter than represented by competing sales representation in the next six months.

Five More Airlines in Staff

In the lower tier, the solo agencies will keep a demonstration airplane available, together with one or more other aircraft for immediate delivery. It is the intent of the company to have a factory demonstration aircraft, will be placed in the demonstration periodically, and made available to the local dealer for demonstration. All these agencies will be subordinate to sell planes on the firm's payment plan, and will provide prompt delivery service arrangements.

The plan calls for the immediate appointment of 100 airframe salesmen and their subordinates in the states.

Notes said that the plan had been made feasible because for the first time since Lockheed's flight, the company is able to make reasonably prompt delivery on planes, and because of the production of the new 100-seat aircraft, the "Skylane," which is to be delivered in January.

Because of the aircraft's size, it makes necessary a cabin-wide aisle staff.

"Selling airplanes today requires

Irving West Coast Plant Nearly Ready



The new manager of the Irving Air Charters Co. at General Control Air Terminal, Glendale, Calif., will be in operation before Jan. 1 in the general engineering office for the maintenance and testing of Irvin Air Charts and accessories. The plant will employ about 250 persons under the management of Guy Hall. Bert White is Western sales manager.

HIGH POINTS
in the
NEWS

* Air asked for operators. McLeod estimates half of those asking aero loans need not to exceed \$100,000,000 for use of individuals or corporations engaged in air transport. Referred to as "airline operators," he means commercial and foreign companies.

* And greater governmental importance now. McLeod also says establishment of a Department of Aeronautics headed by a cabinet member.

* Insurance rule clarified. Barber & Sheldon draft code for guidance of underwriters in gathering aviation insurance statistics.

* Anotherogenesis underway. Hawaiian gets new glider school at 2 hr. 40 min. 100 per day, best effort of American planers. American planes.

* Successful display continues. Baldwin's first supersonic aircraft, having been flown by name forty thousand persons.

* Major aircraft engine. Pratt & Whitney's new engine reported to be 500,000 lbs per hour, best effort of American planers.

* Acquisition of Los Angeles airports' sales of used planes exceeding \$100,000 is under way July 1.

* All airline duty should be standard. So believe James Young, president of Caribbean Print & Whitney, who asks Caribbean Advisory Board for more reliable and prompt delivery on new aircraft models reported to be 100,000 lbs. from same foreign country.

* Acquisition of 100,000 lbs. by Los Angeles airports' sales of used planes July 1. Sixty-five exhibitors already signed.

* And now a touring-demonstrating firm, Curtis Wright. Exhibition Co. will be forced to visit airports, display planes.

* Extensive rail service for Southwest. Three new lines planned. Los Angeles-Bakersfield, Los Angeles-Bakersfield, Denver and Fort Worth-Dallas, and New Orleans-St. Louis. Anticipated as low as between 100 Pass and Fort Worth. And a fourth new line, New York-St. Louis.

* A drop-in, there were two less losses than Air Mail flight for November in 648,000 lbs. to 78,456 lbs. than October.

* Pan American's new Douglas 317 London-Paris flight on a six-plane \$125 for 46 hours service to Paris.

* A formal landing crisis? Well, if the maker will repair it again—that is, if the maker is DeHavilland and if the engine is a Gipsy. Odds good for year after delivery.

Baltimore Show Viewed by 40,000

Layout of Expansion Especially Attractive

BALTIMORE (con't)—Having attracted more than 40,000 persons, the Maryland State Fair, at Baltimore, held at the Pimlico Racecourse, according to an end Saturday evening Dec. 16. Ray Ekins, show director, pronounced the exposition, which had attracted twenty planes, a success and said that a display should be exhibited each year.

The layout of the Baltimore show was outstanding in attractiveness. The high-speed setting, which was completely covered with bright, colorful, strong, and just displays, gave an impression of expansion which was reflected in the show. The visitors, who had experienced long-cold and crowded shows, were surprised at the size of the show and the variety of exhibits. The airplane show, which had covered a large display, was an outstanding feature.

Three Started in Motor

Among the early break exhibits were several which represented products permanent to the aircraft industry, but these were not felt to be as convincing as to deserve the name of the show. The most outstanding exhibit was the new terminal airmail. It had "Edinburgh, Charlotte, Godal, all started on Hatley Division" mattocks!

Showing a host of additional exhibits

Managers of shows thought right well upon the example set at Baltimore, in providing adequate means of transportation for the visitors. The most popular road display by Bob Hall played during the evening and was visible in all parts of the hall because of its central location. Radio music was brought in an area when Hall's men were not on stage.

It is noteworthy that several of the craft exhibited represented firms engaged in business other than aeronautics, including the display of a number of oil and gas business. The Curtis-Robin planes of the American Locomotive and American-Kent organizations were on display as well as the Travel Air monoplane operated by Black and Becker.

Travel Air "W" 20 Seats

Two airplanes, the Travel Air four-seater monoplane and the Travel Air three-seater, were exhibited for the first time at any aircraft show and one, the Soviet Marutec S-26, was displayed for the first time in the East. The Travel Air plane will be presented in a complete performance at the Cleveland Air Races with Dec. 28. This is the first product of the Berlin-Juarez company and probably, an attempt to gain increased visibility in radio competition by using the name of the Soviet Marutec plane so that it is possible to look alike or below it. The Soviet Marutec is a three-place biplane monoplane

More Gibbons, Inc., to Detroit

ST. LOUIS (con't)—The factory of Gibbons, Inc., separated by Detroit Air Corp., located in the St. Louis area, will be moved to a new plant to be completed early in 1939. The new plant will be located in the St. Louis suburb of Webster Groves, Mo., and Gibbons, Inc., will be the largest producer of Southern California owners of Latiflow engines.

has won in production by the American Aeromarine Corp. A Wright company engine is used in the Travel Air plane and Kinner power plants are employed in the R-1 and S-8B craft.

CLEVELAND (con't)—The case of Standard vs Curtis Aeroplane Corp., et al, which was tried before Judge George F. Hall, Dec. 16-17 in Cleveland, is considered of the most interesting inasmuch as it also involved a recent development of the du Pont company. According to the complaint, plaintiffs were a Travel Air lessee, Curtis owners, Fairchild, Kinner, Kinner planes with C-65, Challenger, and J-4 engines; an American Cirrus, and a Gipsy Moth, all of which had a deposable propeller. The aircraft were the Belfins "Prairie Star" and Douglas D-2 and a J-65 Robin, operated by The Travelers Company. Rafts tickets in S-20 were being sold for a Manning place in the state of Montana. The Loring Field 4-Curtis Flying Corps, consolidated O-17 of the Maryland State Guard, and a Curtis biplane.

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Cleveland Court Hears Sweetland vs. Curtiss Case

CLEVELAND (con't)—The case of Standard vs Curtis Aeroplane Corp., et al, which was tried before Judge George F. Hall, Dec. 16-17 in Cleveland, is considered of the most interesting inasmuch as it also involved a recent development of the du Pont company. According to the complaint, plaintiffs were a Travel Air lessee, Curtis owners, Fairchild, Kinner, Kinner planes with C-65, Challenger, and J-4 engines; an American Cirrus, and a Gipsy Moth, all of which had a deposable propeller. The aircraft were the Belfins "Prairie Star" and Douglas D-2 and a J-65 Robin, operated by The Travelers Company. Rafts tickets in S-20 were being sold for a Manning place in the state of Montana. The Loring Field 4-Curtis Flying Corps, consolidated O-17 of the Maryland State Guard, and a Curtis biplane.

Three Started in Motor

Among the early break exhibits were several which represented products permanent to the aircraft industry, but these were not felt to be as convincing as to deserve the name of the show. The most outstanding exhibit was the new terminal airmail. It had "Edinburgh, Charlotte, Godal, all started on Hatley Division" mattocks!

Showing a host of additional exhibits

Managers of shows thought right well upon the example set at Baltimore, in providing adequate means of transportation for the visitors. The most popular road display by Bob Hall played during the evening and was visible in all parts of the hall because of its central location. Radio music was brought in an area when Hall's men were not on stage.

It is noteworthy that several of the craft exhibited represented firms engaged in business other than aeronautics, including the display of a number of oil and gas business. The Curtis-Robin planes of the American Locomotive and American-Kent organizations were on display as well as the Travel Air monoplane operated by Black and Becker.

Travel Air "W" 20 Seats

Two airplanes, the Travel Air four-seater monoplane and the Travel Air three-seater, were exhibited for the first time at any aircraft show and one, the Soviet Marutec S-26, was displayed for the first time in the East. The Travel Air plane will be presented in a complete performance at the Cleveland Air Races with Dec. 28. This is the first product of the Berlin-Juarez company and probably, an attempt to gain increased visibility in radio competition by using the name of the Soviet Marutec plane so that it is possible to look alike or below it. The Soviet Marutec is a three-place biplane monoplane

More Gibbons, Inc., to Detroit

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AVIATION December 22, 1938

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Canadian P & W Weeks Change in Import Duties

OTTAWA (con't)—In a recent appeal made to the Tariff Advisory Board J. A. D. McCord, and James Young, president and managing director of P. W. & W. Co., asked for an increase on the drawback which is a reduction in import duty given for articles already imported into the United States from non-free foreign country.

Mr. McCord said that all import articles should be subject to one set of duties annually, 15 per cent (British) per annum, 25 per cent intermediate and 25 per cent general tariff. He pointed out that the 15 per cent general tariff of 20 per cent could be avoided by the Dominion government on materials or parts required for construction of airplanes provided 40 per cent of the production cost was paid in the United States.

Mr. Young engaged in this industry, he said, felt that the content requirement could not be complied with for at least two years. He urged that drawback be 30 per cent on the intermediate and 25 per cent on the general tariff and stated that he was not in favor of the content requirement being applied right away. The committee accepted the 15 per cent tariff and was to comply with use of a countervailing percentage.

Technical Committee Of U.A.T. in Conference

SEATTLE, WASH.—With engineers and technical experts from various parts of the country in attendance, the annual meeting of the United Aircraft and Transport Corp. has been holding sessions at offices of the Boeing Airplane Co. in this city, with other Boeing companies. In a voluntary of United Aircraft, Boeing, and a group of local directors and P. B. Johnson, president.

During the sessions, which continued for several days, technical details of design and construction of aircraft and aircraft accessories, including monoplane, biplane and engine, commercial air transport operations, dirigible flying schools, and repair services.

The conference was presided over by George J. Moad of Hartford, Conn., chairman of the technical advisory committee, vice-president of Pratt & Whitney, and a director of United.

Ten Campuses for U.S.S.R.

NEW YORK (con't)—Through the Allard Training Corp., formerly Curtis Aeroplane Co., has been ordered for shipment to the U.S.S.R. early in 1939. According to John S. Allard, vice-president and general manager of the Curtis Aeroplane & Motor Co., the engines with spare parts, will represent more than \$100,000.

Naval Squadron Which Won Herbert Schiff Trophy



WASHINGTON (con't)—President Roosevelt on Dec. 14, presented the Herbert Schiff Trophy to Capt. Thomas Garry Fisher on behalf of Training Squadron Seven of the Naval Air Station at San Diego, Calif., which is the Canadian Naval Squadron. Capt. Fisher, a Naval aviator who had flown the greatest number of hours during the year without an accident, a change in the awarding of the trophy was made, as the trophy was to be given to the Canadian Naval Squadron, which had won the 1937 trophy, and the Canadian Naval Squadron was qualified for further training. The officers and eight enlisted men comprising the personnel of the Canadian Naval Squadron, which had won the 1937 trophy, were invited to the ceremony of the presentation of the trophy.

Training Squadron Seven, led by Capt. Fisher with a score of 1,004, this being the highest score ever made by the Canadian Naval Squadron, had charge of the Canadian Naval Squadron, Capt. Fisher, had charge of the Canadian Naval Squadron, one of whose members was Capt. James E. Dye who was the Schiff award on 1938 with a total of 1,257.5 hr. to his credit.

Desination Flies Adds Branch

Bear Repair Station Inspection Problems

WASHINGTON (con't)—Through a four day conference recently completed here, inspection plans were made for the inspection of aircraft repair stations by engineers and inspectors of the Aeromarine Board, and other agencies according to J. S. Marquet, chief of the inspection division of the Aeromarine Board.

A meeting with manufacturers to discuss certain changes suggested will be held in the near future, and will probably be put into effect and will probably be held sometime in the spring. Prior to this, there will be another gathering of the engineers, inspectors and aeromarines to discuss the inspection of aircraft repair stations.

If repair stations comply with the specifications in existing data, which would permit regions to be based on original specifications, it will not be necessary for the stations to send parts and other information to the Aeromarine Board to be done by the Department for approval.

Lanley Model to Byrd and Manly

WASHINGTON (con't)—Captain Richard E. Byrd has been shown to receive the Lanley Model, held for service by the Aeromarine Board, in recognition of his services to the Aeromarine Board and has been notified of the honor through a special radio message to Little America at the annual meeting of the board of regents, president of the Aeromarine Board, Capt. George C. Lanley, and Capt. Charles Marquet, chief of the inspection division of the Aeromarine Board.

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AVIATION
December 11, 1939

BRIEFLY

DeGolasco and Chabenguen has been awarded to Compagnie Aviation Franco-Canadienne.

Proposals for creating a 200-B, major, transport aircraft to be built for \$1,100,000, have been submitted which is to occupy the site of the Walther-Aeroplane in New York City are being considered by officials of the Navy Department.

Construction of four additional aircraft carriers for the Navy have been recommended by Senator Adams.

Progress in the program of the first-line Army aviation program will be held beginning Jan. 7 by the House of Representatives Committee on Military Affairs.

The U.S. Loupian has arrived in Tasmania. After 30 days the oil will help to supply the city with electric power, due to the failure of the city's hydro-electric system because of drought.

An international team for a 1000 km race will be held December 10 in connection with the National Air Races, according to a report of a press committee.

The U.S. coast committee changes its basic race to a 1000 km race of the same in 1940.

A new aeroporto landing gear has been successfully tested and approved by the Department of Commerce.

Only three per cent of passengers on T.A.T.'s Miami-Honolulu flight of 10,000 miles in November, as compared with seven per cent last July.

Plans for a 300-seat refueling flight from New York to Buenos Aires have been announced by Herbert G. Post, president of the company to be made next month.

Only round-trip service between Seattle and Yakima, Wash., has been announced by Northwest Air Express Company.

For the Tampa, Fla., municipal airport, located in a low-lying area, a \$100,000

an automatic device to enable planes to pack up and refuel as flight is directed by J. P. Jacobs, will be built by Southwest Air Transport at Fort Worth, Tex.

Western Air Express is to move its director's office and repair bays to San Antonio City or Los Angeles during the Christmas holidays.

Reports from Washington indicate that the U.S. Shipping Board will refuse to lease ship tonnage to the city of Philadelphia at a nominal cost for port expansion.

A branch office of Detroit Aircraft Corporation is to be established at Pueblo, Colo., according to a recent announcement by Edward S. Evans, president.

Pittsburgh Service & Bldg Company has begun production of Dick aerotowized pianoles, but volume production will not be started until next year.

Canadian Airways, Ltd., has inaugurated daily air and surface between Montreal and the Maritime Provinces.

Plans for carrying mails between

surfboard motorboat for use in case of forced landing on the water nearby.

United States Airways, Inc., plans to begin operation of a passenger line between Wichita, Kan., and Oklahoma City, about January, with Ford as an associate.

Mr. Jim Yule Should have announced a transcontinental solo flight from San Diego to Tucson, is a Shareman with Wright J-5 engine in 30 hr. flying time.

Wright Aeronautical, Inc., Kansas City, Mo., recently accepted a \$100,000 order for 1000 aircraft engines.

Plane owners have found more than \$0 in Saksbury aeroclubs with aircraft gold pins from the company. Merchants will receive similar pins.

Plans of Pacific Air Transport Company are now following stages of completion to put into effect a new trans-Pacific air line between San Francisco and Honolulu, Calif., taking three to 25 hr. air journey.

The Aviation Corporation reports a net loss of \$385,889 for the quarter ended Sept. 30, after depreciation and amortization charges.

General Air Lines, Inc., will soon expand passenger service from San Angelo to San Antonio and Houston. The company has made a 25 per cent reduction of fares during the fall and spring.

Weather has been made to be unpredictable in many parts of the country. Lines operating from Kansas City, Chicago, Milwaukee, and in various parts of Canada have selected several dates during the past two weeks.

Planes have been made to be unpredictable in many parts of the United States.

Transocean Aeronautical Corporation plans to open 100 of its mail planes with radio.

Crescent Air Lines, San Antonio, Tex., reports 30,000 cu. ft. of flying by emergency planes during November.

Pioneer Aircraft Company, Oshkosh, Wis., plans to lease several schools at Falls City and Racine.

Catalin Airways planes flew 90,148 miles during November, carrying 32,484 cu. ft. of mail and 492 passengers.

The Post Office Department is considering making Rockford, Ill., a stop on the air mail route between Chicago and the twin cities.

At a recent 250-mi. test flight from Cleveland to St. Louis, II., the Lockheed twin-engine transoceanic mailplane made radio communication with radio stations between Army planes and the ground station at Clinton Field.

Plans for building a factory and manufacturing surplus are being made by W. E. Biscoe, Vice Manufacturing Company, Joseph Biscoe, and others at Franklin, Pa.

Western Canada Airways, Ltd., has purchased a twin-engine passenger liner for the Winnipeg-Banff service.

Florida headquarters for Sunbeam Aircraft Corporation have been established at Daytona Beach by J. S. Moulton and J. Raymond, who have joined the organization as a branch of United Air Schools there.

Sales in the southeast have been directed by Paul Amerson. Petroleum Company is in control of use of a Sikorsky helicopter for advertising purposes according to company officials.

See 18-45 All aircraft Air Mail, Miami, Fla.

See 18-45 All aircraft Air Mail, Miami, Fla.

AVIATION
December 11, 1939

■■■ PERSONNEL ■■■

SUMMER SMALL, recently general traffic manager of Colony Airways, Inc., has been appointed traffic manager of The Aviation Corporation of America.

PAULUS SEPPONI, traffic manager for

Colonel W. H. Sawyer's replaces

SIDE SLIPS

By
Robert R. O'Brien

THE DEPRESSED AVIATOR stopped by the desk just now for his weekly supply of cigar and in the course of his visit said this: "I saw a clipping from a local paper concerning the proposed flight of the Graf Zeppelin to the North Pole next month."

"Upon the arrival of Captain R. A. Lehman, the Graf Zeppelin with the twelve scientists, will carry fresh provisions for five days—the longest period it is expected to remain away from its bases at Tromsø, Norway, and Fairbanks, Alaska. Emergency provisions for many days will be carried and full equipment for traveling on land or on ice. Twenty-three sledging dogs will go along."

The Journal Aviator says that before he went on that expedition he would wear a clear understanding whether rescue dogs were classified as food provisions or emergency provisions.

Mr. R. A. G. of Lyndhurst, N. J., sends in some thoughts about the department store Santa Claus in White Horse, Peconic, who jumped with a parachute from a airplane over the city. He intended to land in a city park where thousands of children were waiting for him, but missed badly and fell into the Susquehanna River, from which he was rescued by a policeman.

It is a good idea, Mr. G. says, to advise we were much younger than Santa Claus was very unnecessary to his landing of his back-pac, and now it seems he is just as inaccurate in his work with the next pack.

Locum, N.Y.G. of the Polo Alto School of Aviation Polo Alto, Cal., sends in the following copy of a letter of inquiry received by his school:

"Dear Sir—May I ask you whether your school will permit a man who wishes to complete his Primary course, by doing some short work for your school, of course for some others course, the flying which I am willing to do, will be of great value to you, which is very expensive. Trusting you will consider my application as so reasonable. Very truly yours,

"In my opinion clearly a case of some one who has become too "air minded."

OUR HARVEST FLYING
DEPARTMENT

Mr. H. L. C. of Garden City, N. Y., wrote in the office the other day with a good story about an absent-minded pilot, which story was told to him by F. R. of St. Louis. The short-sounding subject of this story is a Navy pilot lost at the time of this incident had been perfect in practically all of his time in land planes. He was transferred to the San Diego station and was again sent to complete his flying for solo use. There was given a Martin MB-2 biplane and the first solo flight about as hard as his flight started so set the ship down on the hard field. His cockpit took the ship away from him, forced it off and made a landing in the water. The absent-minded one was extremely apologetic, explaining that he had had to make time for landmarks that he could always depend on, when he had to set out a map. Then he clouted over the side and dropped overboard.

An Oakland California man is taking a degree on the grounds that his son is "stupid, mean, irritable, morose, disagreeable, mean, grumpy, cool, bitter, judgmental, headstrong, pernicious, judgmental, causing bizarre, mainly, ridiculous, experimental adults, cold, temperamental, immature, unkind and inconsiderate."

This portion of the vocabulary must have been the author of most of the senseless drivel along some new line, and after perusal of some new or plane, we have read mostly:

We are very much pleased with the progress made by the Society for the Amusement Education of Newspaper Correspondents and Artists. The society, of which we are and the president and only member, was started on an educational campaign by the hand of the Society for the Advancement of Photography which ends their appearance in the news columns and responses after Commander Byrd's North Pole flight. They and Bennett were pictured flying to the Pole in everything between

a 1906 engined Jenny with one cockpit part forward of the tail to a Sea Gull flying boat fitted with training board skis.

The various news accounts were equally bad, reporting the Committee on Flying as the main committee of the ship to the tail to make accurate observations, and Bennett up-overhauling an outboard motor which had developed serious oil leaks. In the interests of the younger generation, which might grow up thinking that airplanes really look like those planes we took up the fight and deserved this valuable space for it, these many years, in the recreational efforts of the Society for the Advancement of Photography and its various interests. The work has been hard and disreputable. Many times just as we thought some progress was being made, some paper would print a picture of a ship spinning tail down, or a cogitate advertisement would show the gun house max, of a ship being re-fueled, needless nonchalance on the part of the ship.

Now Mr. Conroy had flown to another Pole and with great fan and trembling we watched the paper. The results were absolutely disastrous. The only serious social error which we have found as far as called to our attention by E. A. H. is the San Angelo, Texas, Evening Standard, which shows a picture of a Pader U-boat in the "long-trimmed position" as it was in the fight. Another disappearance, however, of the "fins" can't be blamed on anyone but G. it is disappearing, nevertheless. On Mr. North Pole flight Byrd saw "no life signs," according the announced New York Times. We have read many of the reports of this flight, appearing in the Times but none of them, to the best of our knowledge, at the South Pole. All of these parts we have been writing with some anxiety for this information, and the Times chooses to overlook that point.

Well, we just accomplish everything in one lifetime, and in general we consider that the work for which the society was organized has been satisfactorily accomplished. As soon as we have digests of the odds and ends of the news columns we will revise to function. Anyone willing to contribute to the fund, however, may make out personal checks to the editor of this paper, and the money will be put to good use immediately.

PARKS TRAINER

STANDARD TRAINING PLANE AT
AMERICA'S LARGEST AIR COLLEGE

Standard training plane at the Parks Air College—largest and best in America—the Parks Trainer is now available to independent flying school operators and to all sportsmen. Especially adapted to student instruction—ideally designed for private pilots—it is the foremost plane of its class on the market today.

Built by a division of the Detroit Aircraft Corporation, the Parks Trainer is backed by an organization which holds a position of acknowledged leadership in the industry. In design and performance it is recognized today as the most satisfactory type of training and sport plane.

Such commanding features as inherent stability and surprising rate of recovery—extremely low landing speed and 50 m. p. h. crossing speed—coupled with positive flight control are a few of the many superiorities of the Parks Trainer.

The Parks Trainer—equipped with an OX-5 engine—is priced at \$3,165. For greater horsepower requirements, the Parks is fitted with an Anzani 159 h. p. engine at \$6,000 and with a Wright J-6, 165 h. p. at \$6,350. Our illustrated folder containing complete details will gladly be sent upon request.

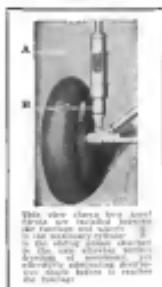
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DETROIT AIRCRAFT CORPORATION

DIVISION OF

Detroit, Michigan

Union Trust Building, Detroit



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The conditions under which Aerol Struts are manufactured make for proven Quality and Performance. The steel going into them is rigidly tested in our modern metallurgical laboratory. Electrically controlled heat treating eliminates the chance of human error. Machined with exacting tolerances and assembled by men who have a tradition of 25 years of building fine products, it is inevitable that every Aerol Strut maintain the highest standard of quality.

In addition to this rigid production control, Aerol Struts are subjected to thorough test in actual operation on our own ships. We have always maintained this policy, believing that no aircraft product can be considered efficient until it has proven itself in the air.

Aerol Struts are manufactured at Cleveland by The Cleveland Pneumatic Tool Company. There is a type and size for every airplane.



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CHUTE*
**Assures Happy
Landings**

THE point of a bow can snap the parabolic out of its pack in record time because this miniature chute is ribbed with spring steel and packed under tension. You jerk the pull ring...it springs out quickly...the big chute fills with air, and you are in flight...gliding gently to earth.

The Irvan Air Chest is available in seat, lap or back pack types. All Irvans are identical in construction and are made in two grades of fine silk, one priced at \$32.00, the other at \$29.00. Every case, regardless of price, complies with the standard U. S. Government packette drawings.

Irvin Air Charter are available in all sections of the country.

Among the important distributors are Curtis Flying Service, Inc., The National Flying Schools, Air Associates, Inc., and Nicholas-Beailey Airplane Co.

Dealers who are interested should communicate directly with the company. If there are no dealers near you, write to us and we will arrange the most convenient way to supply your needs.



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IRVIN *The Life Preserver of the AIR*

IRVING AIR CHUTE CO., INC., 311 PEARL STREET, BUFFALO, N. Y.

ANOTHER ONE OF THE 65 MANUFACTURERS IN THE AVIATION INDUSTRY THAT USES **SKF** BEARINGS
Boeing Air Transport, Inc.



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IN the building of these great transports no expense was spared, and the first requirement of the Boeing Company in purchasing material was... quality. These ships, representing an investment of approximately \$70,000 each and entrusted with the safe transportation of passengers and valuable mail... are equipped with **SKF** Anti-Friction Bearings. That is the story, as told by Boeing Air Transport, Inc., of the part **SKF** products play in the successful operation of their transcontinental air liners. It speaks for itself.

Twenty hours from the Pacific Coast to the Great Lakes! That is the time

these tri-motored giants of the air make, carrying 18 passengers. Each plane has a wing span of 80 feet, length over all of 55 feet, and weighs 21/2 tons when fully loaded. Flying with the largest loads carried anywhere in the United States commercially, over the greatest range in altitude from sea level to 12,000 feet, and temperatures from 30 below zero to 135 degrees above, there could be no compromise with safety and dependability. And **SKF** are making good... to the tune of 10,000 miles of flying every day on the Boeing system.

SKF INDUSTRIES, INC., 40 East 34th Street, New York, N. Y.

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SKF

Ball and Roller Bearings

This Battery is built for hard Flying Service



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Millions of miles of hard flying service stand witness to the rugged qualities of these batteries. Specially designed by skilled engineers, they are planned to meet all the rigid requirements set for air equipment.

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THE ELECTRIC STORAGE BATTERY COMPANY, Philadelphia

Exide Batteries of Canada, Limited, Toronto

NEW TRAFFIC LEAPS ALOFT AS COSTS COME DOWN!



ILYING, today, is a commonplace to thousands—a quicker, cleaner, healthier way of getting from here to there. And this increase is due to cuts in operating cost and the knowing use of good equipment.

The National Air Tour—for the trophy offered by Mr. Edsel Ford—gathered multitudes at every Airport and awoke them to the fact that airplanes operate steadily and economically—in good weather and bad just as the calendar gives it. Each plane covered more than 5,000 miles in 16 days . . . on schedule time.

“Whirlwinds” powered over half the planes on Tour this year. They won 1st, 2nd, and 3rd—70 per cent of all awards in fact! Yet this showing, under gruelling service conditions, is no more than an example of what thousands of “Whirlwinds” are doing . . . everywhere . . . every day!

To all men who use the air, Wright can say that with the new “Whirlwinds,” costs are coming down. The “high cost” bogey of flying is fading from the picture. To those who have not yet flown Wright urges that, with a licensed pilot and a licensed plane, they join the “First Flight Club” in confidence and comfort, behind a “Whirlwind” engine.



WRIGHT
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PATERSON, NEW JERSEY

